

(2) The analysis will include an assessment of the projected losses, net income, and pro forma capital levels and regulatory capital ratios and other capital ratios for the covered company and use such analytical techniques that the Board determines are appropriate to identify, measure, and monitor risks of the covered company that may affect the financial stability of the United States.

■ 10. Section 252.45 is amended by revising paragraph (b)(2) to read as follows:

§ 252.45 Data and information required to be submitted in support of the Board's analyses.

(b) Project a company's pre-provision net revenue, losses, provision for loan and lease losses, and net income; and, pro forma capital levels, regulatory capital ratios, and any other capital ratio specified by the Board under the scenarios described in § 252.44(b).

■ 11. Section 252.52 is amended by:
a. Revising paragraph (n); and
b. removing paragraph (t).
The revision to read as follows:

§ 252.52 Definitions.

(n) Regulatory capital ratio means a capital ratio for which the Board established minimum requirements for the company by regulation or order, including the company's tier 1 and supplementary leverage ratios as calculated under 12 CFR part 217, including the deductions required under 12 CFR 248.12, as applicable, and the company's common equity tier 1, tier 1, and total risk-based capital ratios as calculated under 12 CFR part 217, including the deductions required under 12 CFR 248.12 and the transition provisions at 12 CFR 217.1(f)(4) and 12 CFR 217.300, or any successor regulation; except that, the company shall not use the advanced approaches to calculate its regulatory capital ratios.

■ 12. Section 252.53 is amended by revising paragraph (b)(3) to read as follows:

§ 252.53 Applicability.

(b) Transition periods for covered companies subject to the supplementary leverage ratio. Notwithstanding § 252.52(n) of this subpart, only for purposes of the stress test cycle beginning on January 1, 2016, a bank holding company shall not include an

estimate of its supplementary leverage ratio.

■ 13. Section 252.56 is amended by revising paragraphs (a)(2), (b)(2)(i), and (b)(2)(iv) to read as follows:

§ 252.56 Methodologies and practices.

(2) The potential impact on pro forma regulatory capital levels and pro forma capital ratios (including regulatory capital ratios and any other capital ratios specified by the Board), incorporating the effects of any capital actions over the planning horizon and maintenance of an allowance for loan losses appropriate for credit exposures throughout the planning horizon.

(i) Common stock dividends equal to the quarterly average dollar amount of common stock dividends that the company paid in the previous year (that is, the first quarter of the planning horizon and the preceding three calendar quarters) plus common stock dividends attributable to issuances related to expensed employee compensation;

(iv) An assumption of no issuances of common stock or preferred stock, except for issuances related to expensed employee compensation or in connection with a planned merger or acquisition to the extent that the merger or acquisition is reflected in the covered company's pro forma balance sheet estimates.

■ 14. Section 252.58 is amended by revising paragraphs (b)(3)(v), (b)(4), and (c)(2) to read as follows:

§ 252.58 Disclosure of stress test results.

(v) Pro forma regulatory capital ratios and any other capital ratios specified by the Board;

(4) An explanation of the most significant causes for the changes in regulatory capital ratios; and

(2) The disclosure of pro forma regulatory capital ratios and any other capital ratios specified by the Board that is required under paragraph (b) of this section must include the beginning value, ending value, and minimum value of each ratio over the planning horizon.

By order of the Board of Governors of the Federal Reserve System, July 17, 2015.

Margaret McCloskey Shanks, Deputy Secretary of the Board.

[FR Doc. 2015-18038 Filed 7-22-15; 8:45 am]

BILLING CODE P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-2958; Directorate Identifier 2014-NM-248-AD]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for all The Boeing Company Model 787 airplanes. This proposed AD was prompted by the disclosure that the inner diameters of some batches of landing gear pins were not shot peened in accordance with design specifications and need to be replaced. This proposed AD would require inspection for improperly manufactured landing gear pins, and replacement if necessary. We are proposing this AD to detect and correct insufficient shot peening that could lead to stress corrosion cracking and failure of the landing gear pin, and cause landing gear collapse and inability to control the airplane at high speeds on the ground.

DATES: We must receive comments on this proposed AD by September 8, 2015.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- Federal eRulemaking Portal: Go to http://www.regulations.gov. Follow the instructions for submitting comments.
Fax: 202-493-2251.
Mail: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.
Hand Delivery: Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P. O. Box 3707, MC 2H-65, Seattle, WA 98124-2207;

telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2958.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2958; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Melanie Violette, Senior Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6422; fax: 425-917-6590; email: melanie.violette@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2015-2958; Directorate Identifier 2014-

NM-248-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

We have received a report indicating that the inner diameters of some batches of landing gear pins were not shot peened and need to be replaced. On high strength steel parts, shot peening increases fatigue life and reduces the likelihood of stress corrosion cracking. Stress corrosion cracking, if not corrected, could result in failure of the landing gear pin, and consequent landing gear collapse and the inability to control the airplane at high speeds on the ground.

Related Service Information Under 1 CFR Part 51

We reviewed Boeing Alert Service Bulletin B787-81205-SB320022-00, Issue 001, dated November 14, 2014. The service information describes procedures for the inspection for improperly manufactured landing gear pins (parts that were not shot peened), and replacement if necessary. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section of this NPRM.

FAA's Determination

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

Proposed AD Requirements

This proposed AD would require accomplishing the actions specified in the service information described previously, except as discussed under "Differences Between this Proposed AD and the Service Information." Refer to this service information for details on the procedures and compliance times.

Differences Between This Proposed AD and the Service Information

Boeing Alert Service Bulletin B787-81205-SB320022-00, Issue 001, dated November 14, 2014, limits the effectivity in Group 2 to airplanes delivered prior to the publication of Boeing Alert Service Bulletin B787-81205-SB320022-00, Issue 001, dated November 14, 2014. However, this NPRM does not propose to include that limitation. The applicability of this proposed AD includes all The Boeing Company Model 787 airplanes. Because the affected landing gear pins are rotatable parts, we have determined that these parts could later be installed on production airplanes, thereby subjecting those airplanes to the unsafe condition. This difference has been coordinated with Boeing.

Costs of Compliance

We estimate that this proposed AD affects 13 airplanes of U.S. registry. We estimate the following costs to comply with this proposed AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspection	3 work-hours × \$85 per hour = \$255	\$0	\$255	\$3,315

We estimate the following costs to do any necessary replacements that would

be required based on the results of the proposed inspection. We have no way of

determining the number of aircraft that might need these replacements:

ON-CONDITION COSTS

Action	Labor cost	Parts cost	Cost per product
Replacement	Up to 19 work-hours × \$85 per hour = \$1,615.	\$35,569	Up to \$37,184

Explanation of “RC” Steps in Service Information

The FAA worked in conjunction with industry, under the Airworthiness Directive Implementation Aviation Rulemaking Committee (ARC), to enhance the AD system. One enhancement was a new process for annotating which steps in the service information are required for compliance with an AD. Differentiating these steps from other tasks in the service information is expected to improve an owner's/operator's understanding of crucial AD requirements and help provide consistent judgment in AD compliance. The steps identified as RC (required for compliance) in any service information identified previously have a direct effect on detecting, preventing, resolving, or eliminating an identified unsafe condition.

For service information that contains steps that are labeled as Required for Compliance (RC), the following provisions apply: (1) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD, and an AMOC is required for any deviations to RC steps, including substeps and identified figures; and (2) steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

The Boeing Company: Docket No. FAA–2015–2958; Directorate Identifier 2014–NM–248–AD.

(a) Comments Due Date

We must receive comments by September 8, 2015.

(b) Affected ADs

None.

(c) Applicability

This AD applies to all The Boeing Company Model 787 airplanes, certificated in any category.

(d) Subject

Air Transport Association (ATA) of America Code 32, Landing Gear.

(e) Unsafe Condition

This AD was prompted by the disclosure that the inner diameters of some batches of

landing gear pins were not shot peened in accordance with design specifications and need to be replaced. We are issuing this AD to detect and correct insufficient shot peening that could lead to stress corrosion cracking and failure of the landing gear pin, and cause landing gear collapse and inability to control the airplane at high speeds on the ground.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Inspection and Replacement

At the applicable time specified in paragraph 5, “Compliance,” of Boeing Alert Service Bulletin B787–81205–SB320022–00, Issue 001, dated November 14, 2014, do a landing gear pin part number and serial number inspection, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin B787–81205–SB320022–00, dated November 14, 2014. A review of airplane maintenance or delivery records is acceptable in lieu of this inspection if the part number and serial number of the installed landing gear pins can be conclusively determined from that review.

(1) If no part number or serial number is found that matches the list of affected pin numbers: No further action is required by this paragraph at that pin location.

(2) If any part number or serial number is found that matches the list of affected pin numbers: At the applicable time specified in paragraph 5, “Compliance,” of Boeing Alert Service Bulletin B787–81205–SB320022–00, Issue 001, dated November 14, 2014, replace the affected pin with a pin that does not have an affected part number and serial number, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin B787–81205–SB320022–00, Issue 001, dated November 14, 2014.

(h) Parts Installation Prohibition

As of the effective date of this AD, no person may install on any airplane a landing gear pin having an affected part or serial number identified in Boeing Alert Service Bulletin B787–81205–SB320022–00, Issue 001, dated November 14, 2014.

(i) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (j)(1) of this AD. Information may be emailed to: 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair

required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) For service information that contains steps that are labeled as Required for Compliance (RC), the provisions of paragraphs (i)(4)(i) and (i)(4)(ii) apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

(j) Related Information

(1) For more information about this AD, contact Melanie Violette, Senior Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6422; fax: 425-917-6590; email: melanie.violette@faa.gov.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on July 14, 2015.

Suzanne Masterson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015-17955 Filed 7-22-15; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-2568; Directorate Identifier 2014-SW-026-AD]

RIN 2120-AA64

Airworthiness Directives; Airbus Helicopters (Previously Eurocopter France) Helicopters

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede airworthiness directive (AD) 2014-07-52 for certain Airbus Helicopters (previously Eurocopter France) Model AS350B, AS350BA, AS350B1, AS350B2, AS350B3, AS350C, AS350D, AS350D1, AS355E, AS355F, AS355F1, AS355F2, AS355N, and AS355NP helicopters. AD 2014-07-52 currently requires repetitively inspecting certain reinforcement angles of the rear structure to tailboom junction frame (reinforcement angles) for a crack at 10 hour time-in-service (TIS) intervals, repairing any cracked reinforcement angle, and allows an optional repetitive inspection with a 165 hour TIS inspection interval as a terminating action for the 10 hour TIS inspections. This proposed AD would retain the inspection requirements of AD 2014-07-52 and require the inspection of the area around each reinforcement angle screw hole as terminating action to the 10 hour TIS inspections. These proposed actions are intended to detect a crack in the reinforcement angle, which if not corrected, could result in loss of the tailboom and subsequent loss of control of the helicopter.

DATES: We must receive comments on this proposed AD by September 21, 2015.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Docket:* Go to <http://www.regulations.gov>. Follow the online instructions for sending your comments electronically.

- *Fax:* 202-493-2251.

- *Mail:* Send comments to the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590-0001.

- *Hand Delivery:* Deliver to the "Mail" address between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the European Aviation Safety Agency (EASA) AD, the economic evaluation, any comments received and other information. The street address for the Docket Operations Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

For service information identified in this proposed AD, contact Airbus Helicopters, Inc., 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>. You may review service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy., Room 6N-321, Fort Worth, TX 76177.

FOR FURTHER INFORMATION CONTACT:

Robert Grant, Aviation Safety Engineer, Safety Management Group, FAA, 10101 Hillwood Pkwy., Fort Worth, TX 76177; telephone (817) 222-5110; email robert.grant@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to participate in this rulemaking by submitting written comments, data, or views. We also invite comments relating to the economic, environmental, energy, or federalism impacts that might result from adopting the proposals in this document. The most helpful comments reference a specific portion of the proposal, explain the reason for any recommended change, and include supporting data. To ensure the docket does not contain duplicate comments, commenters should send only one copy of written comments, or if comments are filed electronically, commenters should submit only one time.

We will file in the docket all comments that we receive, as well as a report summarizing each substantive public contact with FAA personnel concerning this proposed rulemaking. Before acting on this proposal, we will consider all comments we receive on or before the closing date for comments. We will consider comments filed after the comment period has closed if it is possible to do so without incurring expense or delay. We may change this proposal in light of the comments we receive.

Discussion

On May 21, 2014, we issued AD 2014-07-52, Amendment 39-17858, 79 FR 33054, June 10, 2014) for Airbus Helicopters Model AS350B, AS350BA, AS350B1, AS350B2, AS350B3, AS350C, AS350D, AS350D1, AS355E, AS355F, AS355F1, AS355F2, AS355N, and AS355NP helicopters with Modification (MOD) 07 3215 installed or with a reinforcement angle, part-number (P/N) 350A08.2493.21 or 350A08.2493.23, installed. AD 2014-07-52 requires, for helicopters with 640 or more hours TIS, within 10 hours TIS and thereafter at