

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2015-2464; Directorate Identifier 2014-NM-195-AD]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede Airworthiness Directive (AD) 2013-22-11, which applies to certain The Boeing Company Model 747-400 and -400D series airplanes. AD 2013-22-11 currently requires repetitive inspections to detect cracks in the floor panel attachment fastener holes of certain upper deck floor beam upper chords, repetitive inspections, corrective actions if necessary, and replacement of the upper deck floor beam upper chords. Since we issued AD 2013-22-11, we received a report that certain fastener holes in the upper deck floor beam upper chords may not have been inspected in accordance with AD 2013-22-11. This proposed AD would add additional repetitive inspections for cracks for certain airplanes, and corrective actions if necessary. We are proposing this AD to detect and correct fatigue cracking in certain upper chords of the upper deck floor beam, which could become large and cause the floor beams to become severed and result in rapid decompression or reduced controllability of the airplane.

DATES: We must receive comments on this proposed AD by September 8, 2015.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707,

MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2464.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2464; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT:

Nathan Weigand, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6428; fax: 425-917-6590; email: Nathan.P.Weigand@faa.gov.

SUPPLEMENTARY INFORMATION:**Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2015-2464; Directorate Identifier 2014-NM-195-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

Structural fatigue damage is progressive. It begins as minute cracks,

and those cracks grow under the action of repeated stresses. This can happen because of normal operational conditions and design attributes, or because of isolated situations or incidents such as material defects, poor fabrication quality, or corrosion pits, dings, or scratches. Fatigue damage can occur locally, in small areas or structural design details, or globally. Global fatigue damage is general degradation of large areas of structure with similar structural details and stress levels. Multiple-site damage is global damage that occurs in a large structural element such as a single rivet line of a lap splice joining two large skin panels. Global damage can also occur in multiple elements such as adjacent frames or stringers. Multiple-site-damage and multiple-element-damage cracks are typically too small initially to be reliably detected with normal inspection methods. Without intervention, these cracks will grow, and eventually compromise the structural integrity of the airplane, in a condition known as widespread fatigue damage (WFD). As an airplane ages, WFD will likely occur, and will certainly occur if the airplane is operated long enough without any intervention.

The FAA's WFD final rule (75 FR 69746, November 15, 2010), became effective on January 14, 2011. The WFD rule requires certain actions to prevent structural failure due to WFD throughout the operational life of certain existing transport category airplanes and all of these airplanes that will be certificated in the future. For existing and future airplanes subject to the WFD rule, the rule requires that design approval holders (DAH) establish a limit of validity (LOV) of the engineering data that support the structural maintenance program. Operators affected by the WFD rule may not fly an airplane beyond its LOV, unless an extended LOV is approved.

The WFD rule (75 FR 69746, November 15, 2010), does not require identifying and developing maintenance actions if the DAHs can show that such actions are not necessary to prevent WFD before the airplane reaches the LOV. Many LOVs, however, do depend on accomplishment of future maintenance actions. As stated in the WFD rule, any maintenance actions necessary to reach the LOV will be mandated by airworthiness directives through separate rulemaking actions.

In the context of WFD, this action is necessary to enable DAHs to propose LOVs that allow operators the longest operational lives for their airplanes, and still ensure that WFD will not occur.

This approach allows for an implementation strategy that provides flexibility to DAHs in determining the timing of service information development (with FAA approval), while providing operators with certainty regarding the LOV applicable to their airplanes.

On October 17, 2013, we issued AD 2013-22-11, Amendment 39-17643 (78 FR 66254, November 5, 2013), for certain The Boeing Company Model 747-400 and -400D series airplanes. AD 2013-22-11 requires repetitive inspections to detect cracks in the floor panel attachment fastener holes of the Section 41 upper deck floor beam upper chords, and corrective actions if necessary; repetitive post-repair and post-modification inspections, and corrective actions if necessary; repetitive inspections of Section 44 upper deck floor beam upper chords, and corrective actions if necessary; repetitive post-repair and post-modification inspections, and corrective actions if necessary; and replacement of the upper deck floor beam upper chords. AD 2013-22-11 superseded AD 2009-10-06, Amendment 39-15901 (74 FR 22424, May 13, 2009). AD 2013-22-11 resulted from an evaluation by the design approval holder (DAH) indicating that certain upper chords of the upper deck floor beam are subject to widespread fatigue damage (WFD). We issued AD 2013-22-11 to detect and correct fatigue cracking in certain upper chords of the upper deck floor beam, which could become large and cause the floor beams to become severed and result in rapid decompression or reduced controllability of the airplane.

Actions Since AD 2013-22-11, Amendment 39-17643 (78 FR 66254, November 5, 2013), Was Issued

Since we issued AD 2013-22-11, Amendment 39-17643 (78 FR 66254,

November 5, 2013), an evaluation by the DAH indicated that certain fastener holes in the upper deck floor beam upper chords in Section 41, that were plugged or re-used during the conversion to a Boeing Converted Freighter, may not have been inspected in accordance with the requirements of AD 2013-22-11, because the locations may be hidden and not recognized as inspection locations. We have determined that, for certain airplanes, it is necessary to add additional repetitive inspections for cracks in the Section 41 upper deck floor beam upper chords and repair if necessary.

Related Service Information Under 1 CFR Part 51

We reviewed Boeing Alert Service Bulletin 747-53A2688, Revision 2, dated August 21, 2014. The service information describes procedures for upper deck floor beam upper chord inspection and repair at floor panel attachment fastener holes in section 41 and section 42. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section of this NPRM. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2464.

FAA’s Determination

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

Proposed AD Requirements

Although this proposed AD does not explicitly restate the requirements of AD 2013-22-11, Amendment 39-17643 (78

FR 66254, November 5, 2013), this proposed AD would retain all of the requirements of AD 2013-22-11. Those requirements are referenced in the service information identified previously, which, in turn, is referenced in paragraphs (g) through (k) of this proposed AD. This proposed AD would add new actions. This proposed AD would require accomplishing the actions specified in the service information identified previously, except as discussed under “Differences Between this Proposed AD and the Service Information.” Refer to this service information for information on the procedures and compliance times.

In addition, the phrase “corrective actions” might be used in this proposed AD. “Corrective actions” are actions that correct or address any condition found. Corrective actions in an AD could include, for example, repairs.

Differences Between This Proposed AD and the Service Information

Boeing Alert Service Bulletin 747-53A2688, Revision 2, dated August 21, 2014, specifies to contact the manufacturer for instructions on how to repair certain conditions, but this proposed AD would require repairing those conditions in one of the following ways:

- In accordance with a method that we approve; or
- Using data that meet the certification basis of the airplane, and that have been approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) whom we have authorized to make those findings.

Costs of Compliance

We estimate that this proposed AD affects 84 airplanes of U.S. registry. We estimate the following costs to comply with this proposed AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspection (retained actions from AD 2013-22-11, Amendment 39-17643 (78 FR 66254, November 5, 2013)).	Up to 309 work-hours × \$85 per hour = \$26,265 per inspection cycle.	\$0	Up to \$26,265 per inspection cycle.	Up to \$2,206,260 per inspection cycle.
New Inspections	Up to 241 work-hours × \$85 per hour = \$20,485.	\$0	Up to \$20,485 per inspection cycle.	Up to \$1,720,740 per inspection cycle.

We have received no definitive data that would enable us to provide a cost estimate for the repair or modification specified in this proposed AD.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of

the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII,

Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the proposed regulation:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:
Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2013–22–11, Amendment 39–17643 (78 FR 66254, November 5, 2013), and adding the following new AD:

The Boeing Company: Docket No. FAA–2015–2464; Directorate Identifier 2014–NM–195–AD.

(a) Comments Due Date

The FAA must receive comments on this AD action by September 8, 2015.

(b) Affected ADs

This AD replaces AD 2013–22–11, Amendment 39–17643 (78 FR 66254, November 5, 2013).

(c) Applicability

(1) This AD applies to The Boeing Company Model 747–400 and –400D series airplanes, certificated in any category, as identified in Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014.

(d) Subject

Air Transport Association (ATA) of America Code 53, Fuselage.

(e) Unsafe Condition

This AD was prompted by an evaluation by the design approval holder (DAH) indicating that certain upper chords of the upper deck floor beam are subject to widespread fatigue damage (WFD). This AD was also prompted by reports that certain fastener holes in the upper deck floor beam upper chords in Section 41, may not have been inspected in accordance with AD 2013–22–11, Amendment 39–17643 (78 FR 66254, November 5, 2013). We are issuing this AD to detect and correct fatigue cracking in certain upper chords of the upper deck floor beam, which could become large and cause the floor beams to become severed and result in rapid decompression or reduced controllability of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Section 41—Repetitive Inspections, and Corrective Actions

At the applicable time specified in table 1 of paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, do open hole or surface high frequency eddy current inspections for cracking of the floor panel attachment holes in the upper deck floor beam upper chords, in accordance with Part 1 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014. If any crack is found during any inspection, before further flight, repair in accordance with Part 2 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, or repair using a method approved in accordance with procedures specified in paragraph (o) of this AD. Repeat the inspections thereafter at the applicable time specified in table 1 of paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, until an action specified in paragraph (g)(1) or (g)(2) of this AD is done.

(1) Doing a repair as a hole modification in accordance with Part 2 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by

paragraph (m)(2) of this AD, terminates the inspections required by paragraph (g) of this AD for the modified hole only.

(2) Doing a modification in accordance with Figure 5 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by paragraph (m)(2) of this AD, terminates the inspections required by paragraph (g) of this AD for the modification only.

(h) Section 41—Repetitive Inspection of Repaired or Modified Holes, and Corrective Actions

For airplanes on which a repair specified in Part 2 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688 is done or a modification specified in Figure 5 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688 is done: At the applicable time specified in table 2 of paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by paragraph (m)(3) of this AD, do open hole or surface high frequency eddy current inspections for cracking of repaired or modified floor panel attachment holes in the upper deck floor beam upper chords, in accordance with Part 1 or Part 3, as applicable, of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014. If any crack is found during any inspection required by this paragraph, before further flight, repair using a method approved in accordance with the procedures specified in paragraph (o) of this AD. Repeat the inspections thereafter at the applicable time specified in table 2 of paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014.

(i) Section 44—Repetitive Inspection, and Corrective Actions

For airplanes identified in Group 1 in Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014: At the applicable time specified in table 3 of paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by paragraph (m)(4) of this AD, do open hole or surface high frequency eddy current inspections of the floor panel attachment holes in the upper deck floor beam upper chords, in accordance with Part 4 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014. If any crack is found during any inspection required by this paragraph, before further flight, repair in accordance with Part 5 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by paragraph (m)(2) of this AD. Repeat the inspections thereafter at the applicable time specified in table 3 of paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, until an action specified in paragraph (i)(1) or (i)(2) of this AD is done.

(1) Doing a repair as a hole modification in accordance with Part 5 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by paragraph (m)(2) of this AD, terminates the inspections required by paragraph (i) of this AD for that modified hole only.

(2) Doing a modification in accordance with Figure 21 of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by paragraph (m)(2) of this AD, terminates the inspections required by paragraph (i) of this AD for that modified hole only.

(j) Section 44—Repetitive Inspection of Repaired or Modified Holes, and Corrective Actions

For airplanes identified in Group 1 in Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, on which a repair specified in Part 5 of Boeing Alert Service Bulletin 747–53A2688 is done or a modification specified in Figure 21 of Boeing Alert Service Bulletin 747–53A2688 is done: At the applicable time specified in table 4 of paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as required by paragraph (m)(3) of this AD, do open hole or surface high frequency eddy current inspections of repaired or modified floor panel attachment holes in the upper deck floor beam upper chords, in accordance with Part 4 or Part 6, as applicable, of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014. If any crack is found during any inspection by this paragraph, before further flight, repair using a method approved in accordance with the procedures specified in paragraph (o) of this AD. Repeat the inspections thereafter at the applicable time specified in table 4 of paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014.

(k) Section 41 and 44—Replacement and Post-Replacement Repetitive Inspections

At the applicable time specified in table 5 of paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014: Replace all upper deck floor beam upper chords, in accordance with Part 7 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014. Within 20,000 flight cycles after doing the replacement, do the inspections specified in paragraphs (g) and (i) of this AD, as applicable. Thereafter, repeat the inspections required by paragraphs (g) and (i) of this AD, as applicable, at the times specified in paragraphs (g) and (i) of this AD.

(l) Section 41—Repetitive Inspection of Plugged or Re-Used Holes, and Corrective Actions

For airplanes identified in Group 2 in Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014: At the applicable time specified in table 6 of paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, except as

required by paragraph (m)(1) of this AD, at all plugged or re-used floor panel attachment holes in the affected floor beam upper chords, do a surface high frequency eddy current inspection of the upper deck floor beam upper chords and detailed inspection for cracks on the vertical flange, in accordance with Part 8 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014. If any crack is found during any inspection required by this paragraph, before further flight, repair using a method approved in accordance with the procedures specified in paragraph (o) of this AD. Repeat the inspections thereafter at the applicable time specified in table 6 of paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014.

(m) Exceptions to Service Information Specifications

(1) Where Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, specifies a compliance time “after the Revision 2 date of this service bulletin,” this AD requires compliance within the specified compliance time after the effective date of this AD.

(2) Where Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014; specifies to contact Boeing for certain procedures: Do the specified actions before further flight using a method approved in accordance with the procedures specified in paragraph (o) of this AD.

(3) Where table 2 or table 4 of paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 747–53A2688, Revision 2, dated August 21, 2014, specifies to contact Boeing for inspections and compliance times: Before further flight, contact the Manager, FAA, Seattle Aircraft Certification Office (ACO), for inspections and compliance times and accomplish the inspections at the given times.

(4) Where Boeing Alert Service Bulletin 747–53A2688, Revision 1, dated September 19, 2012, specifies a compliance time “after the Revision 1 date of this service bulletin,” this AD requires compliance within the specified compliance time after December 10, 2013 (the effective date of AD 2013–22–11, Amendment 39–17643 (78 FR 66254, November 5, 2013)).

(n) Credit for Previous Actions

(1) This paragraph restates the requirements of paragraph (o) of AD 2013–22–11, Amendment 39–17643 (78 FR 66254, November 5, 2013), with new paragraph (h). This paragraph provides credit for the actions required by paragraphs (g) and (h) of this AD, if those actions were performed before December 10, 2013 (the effective date of AD 2013–22–11) using Boeing Alert Service Bulletin 747–53A2688, dated August 21, 2008.

(2) This paragraph provides credit for the actions required by paragraphs (g) through (k) of this AD, if those actions were performed before the effective date of this AD using Boeing Alert Service Bulletin 747–53A2688, Revision 1, dated September 19, 2012, which is not incorporated by reference in this AD.

(o) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (p)(1) of this AD. Information may be emailed to: 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) AMOCs approved for AD 2013–22–11, Amendment 39–17643 (78 FR 66254, November 5, 2013), are approved as AMOCs for the corresponding provisions of paragraphs (g) through (k) of this AD.

(p) Related Information

(1) For more information about this AD, contact Nathan Weigand, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue SW., Renton, WA 98057–3356; phone: 425–917–6428; fax: 425–917–6590; email: Nathan.P.Weigand@faa.gov.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P. O. Box 3707, MC 2H–65, Seattle, WA 98124–2207; phone: 206–544–5000, extension 1; fax: 206–766–5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

Issued in Renton, Washington, on July 15, 2015.

Suzanne Masterson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015–17932 Filed 7–22–15; 8:45 am]

BILLING CODE 4910–13–P