The FAA proposes to amend 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

   Authority: 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):


   **(a) Comments Due Date**
   
   We must receive comments by December 28, 2015.

   **(b) Affected ADs**
   
   None.

   **(c) Applicability**
   

   **(d) Subject**
   
   Air Transport Association (ATA) of America Code 26, Fire Protection.

   **(e) Reason**
   
   This AD was prompted by reports of false engine fire warning events, which consequently led to engine in-flight shut down. We are issuing this AD to prevent unnecessary in-flight shutdown of an engine, which could result in reduced controllability of the airplane.

   **(f) Compliance**
   
   Comply with this AD within the compliance times specified, unless already done.

   **(g) Modification of Engine Fire Extinguishing/Detection System**
   
   Within 18 months after the effective date of this AD: Modify the location and routing of the engine fire detection system, in accordance with the Accomplishment Instructions of EADS CASA Service Bulletin SB–235–26–0006, dated July 8, 2014.

   **(h) Other FAA AD Provisions**
   
   The following provisions also apply to this AD:

   1. Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Shahram Daneshmandi, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1112; fax 425–227–1149. Information may be emailed to: 9-ANM–116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

   2. Notifying the Manufacturer: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or EADS CASA’s EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

   **(ii) Related Information**


   2. For service information identified in this AD, contact EADS–CASA, Military Transport Aircraft Division (MTAD), Integrated Customer Services (ICS), Technical Services, Avenida de Aragón 404, 28022 Madrid, Spain; telephone +34 91 585 55 84; fax +34 91 585 55 05; email MTA.TechnicalService@casa.eads.net; Internet http://www.eads.net. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

   Issued in Renton, Washington, on October 30, 2015.

   Michael Kaszycki, Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

   [FR Doc. 2015–28560 Filed 11–10–15; 8:45 am]

   **BILLING CODE 4910–13–P**

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**


**RIN 2120–AA64**

**Airworthiness Directives; Airbus Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for all
Airbus Model A330–200, –200 Freighter, and –300 series airplanes; and all Airbus Model A340–200, –300, –500, and –600 series airplanes. This proposed AD was prompted by a report of blockage of Angle of Attack (AOA) probes during climb, leading to activation of the Alpha Protection (Alpha Prot) while the Mach number increased. This activation could cause a continuous nose-down pitch rate that cannot be stopped with backward sidestick input, even in the full backward position. For certain airplanes, this proposed AD would require replacing certain AOA sensors (probes) with certain new AOA sensors.

For certain other airplanes, this proposed AD would also require inspections and functional heat testing of certain AOA sensors for discrepancies, and replacement if necessary. We are proposing this AD to prevent erroneous AOA information and Alpha Prot activation due to blocked AOA probes, which could result in a continuous nose-down command and consequent loss of control of the airplane.

**DATES:** We must receive comments on this proposed AD by December 28, 2015.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- Hand Delivery: U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Airbus SAS, Airworthiness Office—EAL, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 45 80; email airworthiness.A330-A340@airbus.com; Internet http://www.airbus.com. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

**Examining the AD Docket**

You may examine the AD docket on the Internet at http://www.regulations.gov by searching for and locating Docket No. FAA–2015–4810; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800–647–5527) is in the addresses section. Comments could result in an AD docket shortly after receipt.


**SUPPLEMENTARY INFORMATION:**

**Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the addresses section. Include “Docket No. FAA–2015–4810; Directorate Identifier 2015–NM–090–AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to http://www.regulations.gov, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

**Discussion**

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2015–0134, dated July 8, 2015 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for all Airbus Model A330–200, –200 Freighter, and –300 series airplanes; and Model A340–200, –300, –500, and –600 series airplanes. The MCAI states:

An occurrence was reported where an Airbus A321 aeroplane encountered a blockage of two Angle of Attack (AOA) probes during climb, leading to activation of the Alpha Protection (Alpha Prot) while the Mach number increased. The flight crew managed to regain full control and the flight landed uneventfully. It was determined that the affected AOA probes are also fitted on A330 and A340 aeroplanes.

When Alpha Prot is activated due to blocked AOA probes, the flight control laws order a continuous nose down pitch rate that, in a worst case scenario, cannot be stopped with backward sidestick inputs, even in the full backward position. If the Mach number increases during a nose down order, the AOA value of the Alpha Prot will continue to decrease. As a result, the flight control laws will continue to order a nose down pitch rate, even if the speed is above minimum selectable speed, known as VLS.

This condition, if not corrected, could result in loss of control of the aeroplane.

Investigation results indicated that aeroplanes equipped with certain UTC Aerospace (UTAS, formerly known as Goodrich) AOA sensors, or equipped with certain SEXTANT/THOMSON AOA sensors, appear to have a greater susceptibility to adverse environmental conditions than aeroplanes equipped with the latest Thales AOA sensor, Part Number (P/N) C16291AB, which was designed to improve AOA indication behaviour in heavy rain conditions.

Having determined that replacement of these AOA sensors is necessary to achieve and maintain the required safety level of the aeroplane, EASA issued AD 2015–0089, to require modification of the aeroplanes by replacement of the affected P/N sensors, and, after modification, prohibits (re-) installation of those P/N AOA sensors. That [EASA] AD also required repetitive detailed visual inspections (DVI) and functional heating tests of certain Thales AOA sensors and provided an optional terminating action for those inspections.

Since EASA AD 2015–0089 was issued, based on further analysis results, Airbus issued Operators Information Transmission (OIT) Ref. 999.0017/15 Revision 1, instructing operators to speed up the removal from service of UTAS P/N 0861ED2 AOA sensors.

For the reasons described above, this [EASA] AD retains the requirements of EASA AD 2015–0089, which is superseded, but reduces the compliance times for aeroplanes with UTAS P/N 0861ED2 AOA sensors installed.


**Related Service Information Under 1 CFR Part 51**

Airbus has issued the following service information:

- **Service Bulletin A330–34–3215, Revision 02, dated March 29, 2010.**
- **Service Bulletin A330–34–3228, dated October 7, 2009.**
- **Service Bulletin A330–34–3315, dated March 26, 2015.**
- **Service Bulletin A340–34–4215, Revision 02, dated March 29, 2010.**
We have received no definitive data that would enable us to provide a cost estimate for the on-condition actions specified in this proposed AD.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority. We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promulgating safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**The Proposed Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended] 

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**Airbus**

Docket No. FAA—2015—4810;
Directorate Identifier 2015—NM—090—AD.

(a) Comments Due Date

We must receive comments by December 28, 2015.

(b) Affected ADs

None.

(c) Applicability

This AD applies to the airplanes, certificated in any category, identified in paragraphs (c)(1) and (c)(2) of this AD, all manufacturer serial numbers.


(d) Subject

Air Transport Association (ATA) of America Code 34, Navigation.

(e) Reason

This AD was prompted by a report of blockage of two Angle of Attack (AOA) probes during climb, leading to activation of the Alpha Protection (Alpha Prot) while the Mach number increased. This activation could cause a continuous nose-down pitch rate that cannot be stopped with backward sidestick input, even in the full backward position. We are issuing this AD to prevent erroneous AOA information and Alpha Prot activation due to blocked AOA probes, which could result in a continuous nose-down command and consequent loss of control of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Replacement of Certain UTAS AOA Sensors

For airplanes on which any UTAS AOA sensor having part number (P/N) 0861ED or P/N 0861ED2 is installed: At the applicable time specified in paragraph (b) of this AD, replace all Captain and First Officer AOA sensors (probes) having P/N 0861ED or 0861ED2 with AOA sensors having Thales P/N C16291AB, in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (g)(1), (g)(2), or (g)(3) of this AD.


(b) Compliance Times for the Requirements of Paragraph (g) of This AD

Do the actions required by paragraph (g) of this AD at the applicable time specified in paragraph (h)(1) or (h)(2) of this AD.

(1) For airplanes with AOA sensors having P/N 0861ED2: Within 22 months after the effective date of this AD.

(2) For airplanes with AOA sensors having P/N 0861ED2: Within 7 months after the effective date of this AD.

(i) Replacement of Certain SEXTANT/THOMSON AOA Sensors

For airplanes on which any SEXTANT/THOMSON AOA sensor having P/N 45150320 is installed: Within 22 months after the effective date of this AD, replace all SEXTANT/THOMSON AOA sensors (probes) having P/N 45150320 with AOA sensors having Thales P/N C16291AB, in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (j)(1), (j)(2), or (j)(3) of this AD.

(ii) Replacement of Certain SEXTANT/THOMSON AOA Sensors

For airplanes on which any SEXTANT/THOMSON AOA sensor having P/N 45150320 or UTAS P/N 0861ED or P/N 0861ED2 has been installed on the airplane since date of issuance of the original airworthiness certificate or date of issuance of the original export certificate of airworthiness.

(2) For airplanes with AOA sensors having P/N 45150320, after accomplishing the replacement.

(ii) The actions specified in paragraphs (g), (j)(1), (j)(2), or (j)(3) of this AD are not required, provided that all conditions specified in paragraphs (l)(2)(i), (l)(2)(ii), and (l)(2)(iii) of this AD are met.

(3) For airplanes with AOA sensors having P/N 45150320, after accomplishing the replacement.

(4) For airplanes on which the replacement required by paragraph (i) of this AD has been done: No person may install, on any airplane, a UTAS AOA sensor having P/N 0861ED or P/N 0861ED2, or a SEXTANT/THOMSON AOA sensor having P/N 45150320, after accomplishing the replacement.

(5) For airplanes on which the replacement required by paragraph (g) of this AD has been done: No person may install, on any airplane, a UTAS AOA sensor having P/N 0861ED or P/N 0861ED2, or a SEXTANT/THOMSON AOA sensor having P/N 45150320, after accomplishing the replacement, except that a UTAS AOA sensor having P/N 0861ED may be installed in the standby position of that airplane.

(o) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; the European Aviation Safety Agency (EASA); or Airbus’s EASA Design Organization Approval (DOA).

(2) The installation is accomplished in accordance with airplane modification instructions approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; the EASA; or Airbus’s EASA DOA.

(m) Optional Terminating Modification

Replacement of all Thales AOA sensors having P/N C16291AA with Thales AOA sensors having P/N C16291AB, in accordance with the Accomplishment Instructions of the applicable service information identified in paragraph (m)(1), (m)(2), or (m)(3) of this AD, terminates the repetitive inspections and functional heating tests required by paragraph (j) of this AD.

(n) Parts Installation Prohibitions

(1) For airplanes on which only Thales P/N C16291AA AOA sensors are installed as of the effective date of this AD: No person may install, on any airplane, a Thales AOA sensor having P/N C16291AA after accomplishing the specified modification.

(2) For airplanes on which the modification specified in paragraph (m) of this AD has been done: No person may install, on any airplane, a Thales AOA sensor having P/N C16291AA after accomplishing the specified modification.

(3) For airplanes on which Thales P/N C16291AA or P/N C16291AB AOA sensors are installed as of the effective date of this AD: No person may install, on any airplane, a UTAS AOA sensor having P/N 0861ED or P/N 0861ED2, or a SEXTANT/THOMSON AOA sensor having P/N 45150320, as of the effective date of this AD.

(4) For airplanes on which the replacement required by paragraph (i) of this AD has been done: No person may install, on any airplane, a UTAS AOA sensor having P/N 0861ED or P/N 0861ED2, or a SEXTANT/THOMSON AOA sensor having P/N 45150320, after accomplishing the replacement.

(5) For airplanes on which the replacement required by paragraph (g) of this AD has been done: No person may install, on any airplane, a UTAS AOA sensor having P/N 0861ED or P/N 0861ED2, or a SEXTANT/THOMSON AOA sensor having P/N 45150320, after accomplishing the replacement, except that a UTAS AOA sensor having P/N 0861ED may be installed in the standby position of that airplane.

(2) For service information identified in this AD, contact Airbus SAS, Airworthiness Office—EAL, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 45 80; email airworthiness.A330-A340@airbus.com; Internet http://www.airbus.com. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

Issued in Renton, Washington, on October 30, 2015.

Michael Kaszyczyk,
Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015–28559 Filed 11–10–15; 8:45 am]
BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION
Federal Aviation Administration

14 CFR Part 39

AIRWORTHINESS DIRECTIVES; BAE SYSTEMS (OPERATIONS) LIMITED MODEL BAe 146 SERIES AIRPLANES AND MODEL AVRO 146–RJ SERIES AIRPLANES

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for all BAE Systems (Operations) Limited Model BAe 146 series airplanes and Model Avro 146–RJ series airplanes. This proposed AD was prompted by reports of cracking of the main fitting of the nose landing gear (NLG) and a determination that a new safe-life limitation for affected NLG main fittings has not been mandated. This proposed AD would require replacing affected NLG main fittings that have exceeded the new limit.

Cracks in the NLG Bell Housing are not detectable with the NLG fitted to the aeroplane and are difficult to detect during overhaul without substantial disassembly of the gear.

This condition, if not corrected, could lead to degradation of direction control on the ground or an un-commanded turn to the left and a consequent loss of control of the airplane on the ground, possibly resulting in damage to the airplane and injury to occupants.

DATES: We must receive comments on this proposed AD by December 28, 2015.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

• Federal eRulemaking Portal: Go to http://www.regulations.gov. Follow the instructions for submitting comments.
• Fax: 202–493–2251.


You may examine the AD docket on the Internet at http://www.regulations.gov, or in person in the Docket Enforcement Group, office (telephone 800–647–5527) is in the AD docket shortly after receipt.

EXAMINING THE AD DOCKET

You may examine the AD docket on the Internet at http://www.regulations.gov by searching for and locating Docket No. FAA–2015–4212; or in person at the Docket Enforcement Group, office (telephone 800–647–5527) is in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT:

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the ADDRESSES section. Include “Docket No. FAA–2015–4212; Directorate Identifier 2015–NM–010–AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to http://www.regulations.gov, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2012–0191R1, dated November 6, 2012 (referred to after this as the Mandatory Continuing Airworthiness Information (the MCAI”), to correct an unsafe condition for all BAE Systems (Operations) Limited Model BAe 146 series airplanes and Model Avro 146–RJ series airplanes. The MCAI states:

Several occurrences of the aeroplane’s Nose Landing Gear (NLG) Main Fitting cracking have been reported. Subsequently in different cases, NLG Main Fitting crack lead to collapsed NLG, locked NLG steering and an aeroplane’s un-commanded steering to the left.

Cracks in the NLG Bell Housing are not detectable with the NLG fitted to the aeroplane and are difficult to detect during overhaul without substantial disassembly of the gear.

This condition, if not corrected, could lead to degradation of direction control on the ground or an un-commanded turn to the left and a consequent loss of control of the airplane on the ground, possibly resulting in damage to the airplane and injury to occupants.

Prompted by these findings, BAE Systems (Operations) Ltd issued Inspection Service Bulletin (ISB) 32–186 (hereafter referred to as the ISB) to introduce a new safe life of 16,000 flight cycles (FC) for certain NLG main fittings, having a Part Number (P/N) as identified in Paragraph 1A, tables 1, 2 and 3 of the ISB. To correct this unsafe condition, EASA issued [EASA] AD 2012–0191 to require implementation of the new safe-life limitation for the affected NLG main fittings and replacement of fittings that have already exceeded the new limit.