International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1137; fax 425–227–1149. Information may be emailed to: 9-ANM-116-AMOC-BREQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Contacting the Manufacturer: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or the EASA; or ATR—GIE Avions de Transport Régional’s EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

(n) Related Information


(2) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (o)(3) and (o)(4) of this AD.

(o) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.


(iv) Messier-Bugatti-Dowty Service Bulletin 631–32–216, Revision 1, dated December 17, 2013. Pages 4, 5, and 8 of this service bulletin are the original issue and are dated October 30, 2013.


(3) For service information identified in this AD, contact ATR—GIE Avions de Transport Régional, 1 Allee Pierre Nadaot, 31712 Blagnac Cedex, France; telephone +33 (0) 5 61 93 36 96; fax +33 (0) 5 61 93 44 51; email account.airworthiness@airbus.com; Internet http://www.airbus.com. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221. It is also available on the Internet at http://www.regulations.gov by searching for and locating Docket No. FAA–2015–0251.


SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to all Airbus Model A318, A319, A320, and A321 series airplanes. The NPRM published in the Federal Register on March 5, 2015 (80 FR 11960). The NPRM was prompted by a determination that, in specific flight conditions, the allowable load limits on the vertical tail plane could be reached and possibly exceeded. Exceeding allowable load could result in detachment of the vertical tail plane. The NPRM proposed to require modification of the pin programming flight warning computer (FWC) to activate the stop rudder input warning (SRIW) logic; and an inspection to determine the part numbers of the FWC and the FAC, and replacement of the FWC and FAC if necessary. We are issuing this AD to prevent detachment of the vertical tail plane and consequent loss of control of the airplane.

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2014–0217 R1, dated February 26, 2015 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition. The MCAI states:

During design reviews that were conducted following safety recommendations related to in-service incidents and one accident on another aircraft type, it has been determined that, in specific flight conditions, the allowable load limits on the vertical tail plane could be reached and possibly exceeded. This condition, if not corrected, could lead, in the worst case, to detachment of the
vertical tail plane in flight and consequent loss of the aeroplane.

To prevent such a possibility, Airbus has developed modifications within the flight augmentation computer (FAC) to reduce the vertical tail plane stress and to activate a conditional aural warning within the flight warning computer (FWC) to further protect against pilot induced rudder doubles.

Consequently, EASA issued AD 2014–0217 (AD.easa.europa.eu/aeb/AD_ad_2014_0217.pdf/AD_2014–0217_1.pdf) to require installation and activation of the stop rudder input warning (SRIW) logic. In addition, the [EASA] AD required, prior to or concurrent with modification of an aeroplane with the activation of the SRIW, upgrades of the FAC and FWC to introduce the SRIW logic and SRIW aural capability, respectively. After modification, the [EASA] AD prohibited installation of certain Part Number (P/N) FWC and FAC.

Since that [EASA] AD was issued, an additional previously-published Airbus Service Bulletin (SB) was identified, and a new SB was published, for the concurrent requirement to replace the FAC with a unit that has a P/N as listed in Table 3 of Appendix 1 of the AD. For the reasons described above, this [EASA] AD is revised to amend paragraph (2), adding references to additional Airbus SBs.

In addition, this AD requires, prior to or concurrent with modification of an airplane with the activation of the SRIW, upgrades of the FAC and FWC to introduce the SRIW logic and SRIW aural capability, respectively. After modification, this AD prohibits installation of FWCs and FACs having certain part numbers. You may examine the MCAI in the AD docket on the Internet at http://www.regulations.gov/#/documentDetail;D=FAA-2015-0251-0003.

Comments
We gave the public the opportunity to participate in developing this AD. The following presents the comments received on the NPRM (80 FR 11960, March 5, 2015) and the FAA's response to each comment.

Request To Refer to Revised Service Information

Airbus requested that we refer to revised service information.

We agree with the Airbus request to refer to revised service information. No additional work is required by the revised service information. We have revised paragraph (g) of this AD to refer to Airbus Service Bulletin A320–22–1480, Revision 02, dated March 30, 2015. We have added new paragraph (m)(1) of this AD to provide credit for the actions required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using Airbus Service Bulletin A320–22–1480, dated July 9, 2014; or Airbus Service Bulletin A320–22–1480, Revision 01, dated February 6, 2015.

We have revised paragraph (i) of this AD to refer in part to the following service information.

- Airbus Service Bulletin A320–22–1427, Revision 05, including Appendix 01, dated November 24, 2014 (FWC 622 hard B).
- Airbus Service Bulletin A320–22–1447, Revision 03, dated April 21, 2015 (FAC CAA02 hard C).

We have redesignated paragraph (m) of the proposed AD (80 FR 11960, March 5, 2015) as new paragraph (m)(2) of this AD to provide credit for the actions required by paragraph (i) of this AD, if those actions were performed before the effective date of this AD using the following additional service information.


Request To Clarify Approved Parts

United Airlines (UAL) requested that we split paragraph (h)(3)(iv) of the proposed AD (80 FR 11960, March 5, 2015) into two paragraphs to clarify the approved parts. UAL stated that paragraphs (h)(3)(i), (h)(3)(ii), and (h)(3)(iii) of the proposed AD clearly denote three of the four possible combinations of altitude and speed. UAL stated that (h)(3)(v) of the proposed AD leads one to believe that a FAC CAA02 hard C is required regardless of the airplane configuration.

We agree with UAL’s request to clarify the FACs and FWCs having the part numbers that are compatible with SRIW activation required by paragraph (g) of this AD. We have revised paragraph (h)(3)(iv) of the AD to state that for all airplanes configured with an FAC standard CAA01, an FAC having soft P/N G2856AA02 installed on hard P/N C13206AA00 (CAA02 hard C) are compatible with SRIW activation required by paragraph (g) of this AD. We have added new paragraph (h)(3)(v) of this AD to state that for all airplane configurations, an FWC having P/N 350E053021212 (H2–F7) are compatible with SRIW activation required by paragraph (g) of this AD.
plane (VTP) loads remain within the safe limits. After reviewing the design, analyses, and simulator demonstrations, the FAA has concluded that these warnings will prevent the flightcrew from continuing the inappropriate rudder inputs prior to exceeding the ultimate design loads that could result in failure of the vertical tail plane. We have determined that details associated with our disposition to NTSB safety recommendations A–04–56 and A–04–57 are outside the context of this AD. We will provide those details directly to the NTSB in our response to the safety recommendations. We have not changed this final rule in this regard.

Conclusion
We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting this AD with the changes described previously and minor editorial changes. We have determined that these minor changes:

• Are consistent with the intent that was proposed in the NPRM (80 FR 11960, March 5, 2015) for correcting the unsafe condition; and

• Do not add any additional burden upon the public than was already proposed in the NPRM (80 FR 11960, March 5, 2015).

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

Related Service Information Under 1 CFR Part 51
Airbus has issued Service Bulletin A320–22–1480, Revision 02, dated March 30, 2015. This service information describes procedures for modifying the pin programming to activate the SRIW logic.

Airbus has also issued the following service information. The service information describes procedures for replacing FWCs and FACs.

• Airbus Service Bulletin A320–22–1447, Revision 03, dated April 21, 2015.

This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section of this AD.

Costs of Compliance
We estimate that this AD affects 953 airplanes of U.S. registry.

We also estimate that it will take about 3 work-hours per product to comply with the basic requirements of this AD. The average labor rate is $85 per work-hour. Based on these figures, we estimate the cost of this AD on U.S. operators to be $243,015, or $255 per product.

In addition, we estimate that any necessary follow-on actions will take about 6 work-hours (3 work-hours for an FWC and 3 work-hours for an FAC), for a cost of up to $510 per product. We have received no definitive data that will enable us to provide part cost estimates for the on-condition actions specified in this AD. We have no way of determining the number of aircraft that might need these actions.

Authority for This Rulemaking
Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings
We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a “significant regulatory action” under Executive Order 12866;
(d) Subject
Air Transport Association (ATA) of America Code 22, Auto Flight; 31, Instruments.

(e) Reason
This AD was prompted by a determination that, in specific flight conditions, the allowable load limits on the vertical tail plane could be reached and possibly exceeded. Exceeding allowable load could result in detachment of the vertical tail plane. We are issuing this AD to prevent detachment of the vertical tail plane and consequent loss of control of the airplane.

(f) Compliance
Comply with this AD within the compliance times specified, unless already done.

(g) Pin Programming Modification
Within 48 months after the effective date of this AD, modify the pin programming to activate the stop rudder input warning (SRIW) logic, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320–22–1480, Revision 02, dated March 30, 2015.

(h) Inspection To Determine Part Numbers (P/Ns), Flight Warning Computer (FWC) and Flight Augmentation Computer (FAC) Replacement
Prior to or concurrently with the actions required by paragraph (g) of this AD: Inspect the part numbers of the FWC and the FAC installed on the airplane. If any FWC or FAC having a part number identified in paragraph (h)(1) or (h)(2) of this AD, as applicable, is installed on an airplane, prior to or concurrently with the actions required by paragraph (g) of this AD, replace all affected FWCs and FACs with a unit having a part number identified in paragraph (h)(3) of this AD, in accordance with the Accomplishment Instructions of the applicable Airbus service information specified in paragraph (i) of this AD.

(1) Paragraphs (h)(1)(i) through (h)(1)(viii) of this AD identify FWCs having part numbers that are incompatible with the SRIW activation required by paragraph (g) of this AD.

(i) B397AAM0202.
(ii) B397BAM0101.
(iii) B397BAM0512.
(iv) B397AAM0301.
(v) B397BAM0202.
(vi) B397BAM0513.
(vii) B397AAM0302.
(viii) B397BAM0203.
(ix) B397BAM0514.
(x) B397AAM0303.
(xi) B397BAM0305.
(xii) B397BAM0505.
(xiii) B397AAM0404.
(xiv) B397BAM0406.
(xv) B397BAM0616.
(xvi) B397AAM0405.
(xvii) B397BAM0407.
(xviii) B397BAM0617.
(xix) B397AAM0506.
(xx) B397BAM0507.
(xxi) B397BAM0618.
(xxii) B397AAM0507.
(xxiii) B397BAM0606.
(xxiv) B397BAM0508.
(xxv) B397BAM0509.
(xxvi) B397BAM0620.
(xxvii) B397AAM0509.
(xxviii) B397BAM0510.
(xxix) B397CAM0101.
(xxx) B397BAM0510.
(xxxi) B397BAM0511.
(xxxii) B397CAM0102.

(2) Paragraphs (h)(2)(i) through (h)(2)(xxvii) of this AD identify FACs having part numbers that are non-compatible with the SRIW activation required by paragraph (g) of this AD.

(i) 350E017238484 (H1D1).  
(ii) 350E0503020603 (H2E3).
(iii) 350E016187171 (C3).
(iv) 350E0503020404 (H2E4).
(v) 350E017248685 (H1D2).
(vi) 350E0503020666 (H2F2).
(vii) 350E017251414 (H1E1).
(viii) 350E0503020707 (H2F3).
(ix) 350E017271616 (H1E2).
(x) 350E0503021010 (H2F3P).
(xi) 350E0182911818 (H1E3CJ).
(xii) 350E0503022088 (H2F4).
(xiii) 350E0183091919 (H1E3P).
(xiv) 350E0503020909 (H2F5).
(xv) 350E017216120 (H1E4Q).
(xvi) 350E0503021111 (H2F6).
(xvii) 350E0503020202 (H2E2).

(2) Paragraphs (h)(2)(i) through (h)(2)(xxvii) of this AD identify FACs having part numbers that are non-compatible with the SRIW activation required by paragraph (g) of this AD.

(j) Exclusion From Actions Required by Paragraphs (g) and (h) of This AD
An airplane on which Airbus Modification 154473 has been embodied in production is excluded from the requirements of paragraphs (g) and (h) of this AD, provided that within 30 days after the effective date of this AD, an inspection of the part numbers of the FWC and the FAC installed on the airplane is done to determine that no FWC having a part number listed in paragraph (h)(1) of this AD, and no FAC having a part number listed in paragraph (h)(2) of this AD, has been installed on that airplane since date of manufacture. A review of airplane maintenance records is acceptable in lieu of this inspection if the part numbers of the FWC and FAC can be conclusively determined from that review. If any FWC or FAC having a part number identified in paragraph (h)(1) or (h)(2) of this AD, as applicable, is installed on a post-Airbus Modification 154473 airplane: Within 30 days after the effective date of this AD, do the replacement required by paragraph (h) of this AD.

(k) Parts Installation Prohibitions
After modification of an airplane as required by paragraphs (g), (h), and (j) of this AD: Do not install on that airplane any FWC having a part number listed in paragraph (h)(1) of this AD or any FAC having a part number listed in paragraph (h)(2) of this AD.

(l) Later Approved Parts
Installation of a version (part number) of the FWC or FAC approved after the effective date of this AD is an approved method of compliance with the requirements of paragraph (b) or (j) of this AD; provided the requirements specified in paragraphs (l)(1) and (l)(2) of this AD are met.

(1) The version (part number) must be approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus’s EASA Design Organization Approval (DOA).

(2) The installation must be accomplished using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or EASA; or Airbus’s EASA DOA.

(m) Credit for Previous Actions
(1) This paragraph provides credit for actions required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using Airbus Service Bulletin A320–22–1480, dated July 9, 2014; or Airbus Service Bulletin A320–22–1480, Revision 01, dated February 6, 2015. This service information is not incorporated by reference in this AD.

(2) This paragraph provides credit for actions required by paragraph (i) of this AD, if those actions were performed before the effective date of this AD using the applicable Airbus service information identified in paragraphs [m][2][i] through [m][2][xviii] of
this AD. This service information is not incorporated by reference in this AD.


(n) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, your request to the appropriate inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Sanjay Kalhan, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1405; fax 425–227–1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local Flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Required for Compliance (RC): If any service information contains procedures or tests that are RC, those procedures and tests must be done to comply with this AD; any procedures or tests that are not identified as RC are recommended. Those procedures and tests that are not identified as RC may be deviated from using accepted methods in accordance with the operator’s maintenance or inspection program without obtaining approval of an AMOC, provided the procedures and tests identified as RC can be done and the airplane can be put back in an airworthy condition. Any substitutions or changes to procedures or tests identified as RC require approval of an AMOC.

(3) Contacting the Manufacturer: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or the EASA; or Airbus’s EASA DOA. If approved by the DOA, the approval must include the DOA–authorized signature.

(o) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2014–0217R1, dated February 26, 2015, for related information. This MCAI may be found in the AD docket on the Internet at http://www.regulations.gov/#!documentDetail;D=FAA-2015–0251– 0005

(2) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (p)(3) and (p)(4) of this AD.

(p) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.


(3) For service information identified in this AD, contact Airbus, Airworthiness Office—ELAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet http://www.airbus.com.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: http://www.archives.gov/federal-register/cfr/ibr-locations.html.

Issued in Renton, Washington, on November 9, 2015.

Michael Kaszycki,
Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015–29702 Filed 11–23–15; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 71

[Docket No. FAA–2015–1869; Airspace Docket No. 15–AGL–9]

Establishment of Class E Airspace; Newberry, MI

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule; correction.

SUMMARY: This action corrects an error in the legal description of a final rule published in the Federal Register of September 24, 2015, that establishes Class E airspace at the Newberry VHF Omni-Directional Range/Distance Measuring Equipment (VOR/DME), Newberry, MI. The legal description noted exclusionary language for Federal Airways and Canadian airspace not required for this airspace.

DATES: Effective date: 0901 UTC, December 10, 2015. The Director of the Federal Register approves this incorporation by reference action under 1 Code of Federal Regulations, Part 51, subject to the annual revision of FAA Order 7400.9 and publication of conforming amendments.

FOR FURTHER INFORMATION CONTACT: Raul Garza, Jr., Central Service Center, Operations Support Group, Federal Aviation Administration, Southwest Region, 10101 Hillwood Parkway., Fort Worth, TX 76177; telephone 817–222–5874.

SUPPLEMENTARY INFORMATION:

History

On September 24, 2015, a final rule was published in the Federal Register establishing Class E airspace at the Newberry VOR/DME, Newberry, MI (80 FR 57522) Docket No. FAA–2015–1869. Subsequent to publication, the FAA found that the exclusionary language for Federal Airways and Canadian airspace noted in the airspace description is not required and, therefore, is removed.

Final Rule Correction

Accordingly, pursuant to the authority delegated to me, in the Federal Register of September 24, 2015, (80 FR 57522) FR Doc. 2015–23987, on