

## The Special Conditions

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Cessna Aircraft Company, 208 and 208B Caravan airplanes modified by Peregrine.

### 1. Installation of Lithium Battery

The FAA states in this Notice that the following special conditions be applied to lithium battery installations on the 208 and 208B airplanes in lieu of the requirements § 23.1353(a)(b)(c)(d)(e), amendment 49.

Lithium battery installations on the 208 and 208B airplanes must be designed and installed as follows:

a. Safe cell temperatures and pressures must be maintained during any probable charging or discharging condition, or during any failure of the charging or battery monitoring system not shown to be extremely remote. The lithium battery installation must be designed to preclude explosion or fire in the event of those failures.

b. Lithium batteries must be designed to preclude the occurrence of self-sustaining, uncontrolled increases in temperature or pressure.

c. No explosive or toxic gasses emitted by any lithium battery in normal operation or as the result of any failure of the battery charging or monitoring system, or battery installation not shown to be extremely remote, may accumulate in hazardous quantities within the airplane.

d. Lithium batteries that contain flammable fluids must comply with the flammable fluid fire protection requirements of 14 CFR 23.863(a) through (d).

e. No corrosive fluids or gases that may escape from any lithium battery may damage airplane structure or essential equipment.

f. Each lithium battery installation must have provisions to prevent any hazardous effect on structure or essential systems that may be caused by the maximum amount of heat the battery can generate during a short circuit of the battery or of its individual cells.

g. Lithium battery installations must have—

(1) A system to control the charging rate of the battery automatically to prevent battery overheating or overcharging, or

(2) A battery temperature sensing and over-temperature warning system with a means for automatically disconnecting the battery from its charging source in the event of an over-temperature condition or,

(3) A battery failure sensing and warning system with a means for automatically disconnecting the battery from its charging source in the event of battery failure.

h. Any lithium battery installation functionally required for safe operation of the airplane, must incorporate a monitoring and warning feature that will provide an indication to the appropriate flight crewmembers, whenever the capacity and state of charge of the batteries have fallen below levels considered acceptable for dispatch of the airplane.

i. The ICAW must contain recommended manufacturer's maintenance and inspection requirements to ensure that batteries, including single cells, meet a functionally safe level essential to the aircraft's continued airworthiness.

(1) The ICAW must contain operating instructions and equipment limitations in an installation maintenance manual.

(2) The ICAW must contain installation procedures and limitations in a maintenance manual, sufficient to ensure that cells or batteries, when installed according to the installation procedures, still meet safety functional levels essential to the aircraft's continued airworthiness. The limitations must identify any unique aspects of the installation.

(3) The ICAW must contain corrective maintenance procedures to check battery capacity at manufacturer's recommended inspection intervals.

(4) The ICAW must contain scheduled servicing information to replace batteries at manufacturer's recommended replacement time.

(5) The ICAW must contain maintenance and inspection requirements how to check visually for battery and charger degradation.

j. Batteries in a rotating stock (spares) that have degraded charge retention capability or other damage due to prolonged storage must be checked at manufacturer's recommended inspection intervals.

k. If the lithium battery application contains software and/or complex hardware, in accordance with AC 20-115<sup>1</sup> and AC 20-152,<sup>2</sup> they should be developed to the standards of DO-178 for software and DO-254 for complex hardware.

Compliance with the requirements of this Special Condition must be shown

<sup>1</sup> [http://rgl.faa.gov/Regulatory\\_and\\_Guidance\\_Library/rgAdvisoryCircular.nsf/0/E35FBC0060E2159186257BBE00719FB3?OpenDocument&Highlight=ac%2020-115b](http://rgl.faa.gov/Regulatory_and_Guidance_Library/rgAdvisoryCircular.nsf/0/E35FBC0060E2159186257BBE00719FB3?OpenDocument&Highlight=ac%2020-115b).

<sup>2</sup> [http://rgl.faa.gov/Regulatory\\_and\\_Guidance\\_Library/rgAdvisoryCircular.nsf/0/6D4AE0BF1BDE3579862570360055D119?OpenDocument&Highlight=ac%2020-152](http://rgl.faa.gov/Regulatory_and_Guidance_Library/rgAdvisoryCircular.nsf/0/6D4AE0BF1BDE3579862570360055D119?OpenDocument&Highlight=ac%2020-152).

by test or analysis, with the concurrence of the Wichita Aircraft Certification Office.

Issued in Kansas City, Missouri on May 9, 2016.

**Pat Mullen,**

*Acting Manager, Small Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2016-11502 Filed 5-13-16; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2015-2462; Directorate Identifier 2014-NM-224-AD; Amendment 39-18515; AD 2016-10-04]

**RIN 2120-AA64**

#### Airworthiness Directives; The Boeing Company Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for certain The Boeing Company Model 737-100, -200, -200C, -300, -400, and -500 series airplanes. This AD was prompted by reports of cracked antenna support channels, skin cracking underneath the number 2 very high frequency (VHF) antenna, and cracking in the frames attached to the internal support structure. This AD requires repetitive inspections to determine the condition of the skin and the internal support structure, and follow-on actions including corrective action as necessary. We are issuing this AD to detect and correct skin cracking of the fuselage. Such cracking could result in separation of the number 2 VHF antenna from the airplane and rapid depressurization of the cabin.

**DATES:** This AD is effective June 20, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of June 20, 2016.

**ADDRESSES:** For service information identified in this final rule, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on

the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2462.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2462; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Jennifer Tsakoumakis, Aerospace Engineer, Airframe Branch, ANM-120L, FAA, Los Angeles Aircraft Certification Office (ACO), 3960 Paramount Boulevard, Lakewood, CA 90712-4137; phone: 562-627-5264; fax: 562-627-5210; email: [jennifer.tsakoumakis@faa.gov](mailto:jennifer.tsakoumakis@faa.gov).

#### SUPPLEMENTARY INFORMATION:

##### Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to certain The Boeing Company Model 737-100, -200, -200C, -300, -400, and -500 series airplanes. The NPRM published in the **Federal Register** on July 20, 2015 (80 FR 42756) (“the NPRM”). The NPRM was prompted by reports of cracked antenna support channels, skin cracking underneath the number 2 VHF antenna, and cracking in the frames attached to the internal support structure. The NPRM proposed to require repetitive inspections to determine the condition of the skin and the internal support structure, and follow-on actions including corrective action as necessary. We are issuing this AD to detect and correct skin cracking of the fuselage. Such cracking could result in separation of the number 2 VHF antenna from the airplane and rapid depressurization of the cabin.

##### Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comments

received on the NPRM and the FAA’s response to each comment. Boeing concurred with the NPRM.

#### Request To Correct a Typographical Error in Paragraph (h)(4) of the Proposed AD

Southwest Airlines requested that we correct a typographical error in paragraph (h)(4) of the proposed AD, which states that accomplishment of the preventative modification terminates the inspection required by “paragraphs (g), (g)(1), and (h)(2) of the AD.” Southwest Airlines noted that the NPRM does not contain paragraph (g)(1). Southwest Airlines concluded that this appears to be a typographical error and the references to paragraphs (g) and (g)(1) of the proposed AD should be to paragraphs (h) and (h)(1) of the proposed AD, similar to what is stated in paragraph (k)(3) of the proposed AD.

British Airways stated that it has identified a potential contradiction between paragraphs (h)(4) and (k)(3) of the proposed AD. British Airways stated that paragraph (h)(4) of the proposed AD refers to paragraph (h)(2), whereas paragraph (k)(3) of the proposed AD refers to paragraphs (h), (h)(1), and (h)(2) of the proposed AD.

We agree to revise paragraph (h)(4) of this AD because there is a typographical error. We have changed the references in paragraph (h)(4) of this AD to specify paragraphs (h), (h)(1), and (h)(2) of this AD. This change resolves the contradiction noted by British Airways.

#### Request a Provision To Terminate Inspections Required by Paragraph (h) of the Proposed AD

Southwest Airlines requested that we provide a provision to terminate the inspections required by paragraph (h) of the proposed AD for previously installed repairs that have received FAA approval. The commenter stated that these repairs would inhibit the inspections required by paragraph (h) of this AD.

We do not agree with the commenter’s request because previously installed FAA-approved repairs may not have been designed to address the specified unsafe condition identified in this AD. We understand that some of these repairs may not allow inspection of the area specified in the AD; in those cases, the operator must request approval for an alternative method of compliance (AMOC) according to paragraph (m) of this AD. We have not revised this AD in this regard.

#### Request To Add Terminating Action to Paragraphs (h)(1) and (h)(2) of the Proposed AD

British Airways asked why the terminating action specified in paragraph (k)(3) of the proposed AD is not included in the text “until the accomplishment of paragraphs” references in paragraphs (h)(1) and (h)(2) of the proposed AD. We infer British Airways is requesting that we revise paragraphs (h)(1) and (h)(2) of the proposed AD.

We agree with the commenter because installation of the preventive modification in accordance with paragraph (k)(3) of this AD is acceptable for terminating the repetitive inspections. In addition, we note the reference to paragraph (k)(1) of this AD in paragraph (h)(2) of this AD is redundant. We have made the following changes to this AD:

- In paragraph (h)(1) of this AD, we specify to repeat the inspections “until the accomplishment of paragraph (k)(1), (k)(2), or (k)(3) of this AD, as applicable.”
- In paragraph (h)(2) of this AD, we specify to repeat the inspections “until the accomplishment of paragraph (k)(2) or (k)(3) of this AD, as applicable.”

#### Request To Correct the Language in Paragraph (h)(2) of the Proposed AD

Southwest Airlines requested a correction to the language in paragraph (h)(2) of the proposed AD to add the term “as applicable” after the listed inspections. Southwest Airlines stated that there are multiple sections of Part 2 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737-53-1159, Revision 1, dated October 20, 2014. Southwest Airlines noted that each group/configuration has its own Part 2 instructions and that Groups 3 through 6, Configurations 2 and 3, do not contain instructions for internal detailed inspections or internal high frequency eddy current inspections.

We agree with the commenter because certain inspections are applicable to only certain configurations. We have added the language “as applicable” to paragraph (h)(2) of this AD.

#### Request To Revise Certain Paragraphs To Include a Terminating Action for the Preventive Modification

Southwest Airlines requested that we revise paragraph (k)(2) of the proposed AD to include a statement that accomplishment of the repair specified in paragraph (h)(3) of the proposed AD also terminates the preventive modification specified in paragraph

(h)(4) of the proposed AD. Southwest Airlines stated that there is no language in the NPRM or Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014, that states whether the preventative modification is required after the repair is installed.

We do not agree with the commenter’s request because, for some airplane configurations, the repair only installs an external skin doubler and the preventative modification includes replacement of the internal support structure. For some airplane configurations, the preventive modification specified in Part 4 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014, is required after installation of the repair specified in Part 3 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014. We have not revised this AD in this regard.

**Request To Provide Statement for Terminating Actions in Paragraph (k)(2) of the Proposed AD**

Southwest Airlines requested that we revise paragraph (k)(2) of the proposed AD to specify the repair also terminates the initial inspections in paragraph (h) of the proposed AD. Southwest Airlines stated that the current statement in paragraph (k)(2) of the proposed AD does not address a terminating action for the initial inspection specified in paragraphs (h)(1) and (h)(2) of the proposed AD for aircraft that have previously installed the repair specified in paragraph (h)(3) of the proposed AD.

We agree with the commenter’s request because repairs installed in accordance with Part 3 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014, prior to the effective date of this AD, will not allow accomplishment of the initial inspections as specified in paragraph (h) of this AD. We revised paragraph (k)(2) of this AD to specify that accomplishment of the repair required by paragraph (h)(3) of this AD terminates the initial and repetitive inspections required in paragraphs (h), (h)(1), and (h)(2) of this AD.

**Effect of Winglets on Accomplishment of the Proposed Actions**

Aviation Partners Boeing stated that accomplishing Supplemental Type Certificate (STC) ST01219SE ([http://rgl.faa.gov/Regulatory\\_and\\_Guidance\\_Library/rgstc.nsf/0/be866b732f6cf31086257b9700692796/\\$FILE/ST01219SE.pdf](http://rgl.faa.gov/Regulatory_and_Guidance_Library/rgstc.nsf/0/be866b732f6cf31086257b9700692796/$FILE/ST01219SE.pdf)) does not affect the actions specified in the NPRM.

We concur with the commenter. We have redesignated paragraph (c) of the proposed AD as paragraph (c)(1) of this AD and added new paragraph (c)(2) to this AD to state that installation of STC ST01219SE does not affect the ability to accomplish the actions required by this AD. Therefore, for airplanes on which STC ST01219SE is installed, a “change in product” AMOC approval request is not necessary to comply with the requirements of 14 CFR 39.17.

**Change to Paragraph (j) of This AD**

We have revised paragraph (j) of this AD to clarify that the post-repair and post-modification inspections are airworthiness limitations that are

required by maintenance and operational rules; therefore, these inspections are not required by this AD.

**Conclusion**

We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting this AD with the changes described previously and minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

**Related Service Information Under 1 CFR Part 51**

We reviewed Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014. The service information describes procedures for repetitive inspections to determine the condition of the skin and the internal support structure, and follow-on actions including corrective action as necessary. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

**Costs of Compliance**

We estimate that this AD affects 609 airplanes of U.S. registry.

We estimate the following costs to comply with this AD:

**ESTIMATED COSTS**

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspections .....	33 work-hours × \$85 per hour = \$2,805 per inspection cycle.	\$0	\$2,805 per inspection cycle .....	\$1,708,245 per inspection cycle.

We estimate the following costs to do any necessary repairs/modifications that

would be required based on the results of the inspections. We have no way of

determining the number of aircraft that might need these repairs/modifications.

**ON-CONDITION COSTS**

Action	Labor cost	Parts cost	Cost per product
Repair and preventive modification .....	63 work-hours × \$85 per hour = \$5,355 .....	\$10,432	\$15,787

According to the manufacturer, some of the costs of this AD may be covered under warranty, thereby reducing the cost impact on affected individuals. We

do not control warranty coverage for affected individuals. As a result, we have included all costs in our cost estimate.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA’s authority to issue

rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

### Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**2016–10–04 The Boeing Company:**  
Amendment 39–18515; Docket No. FAA–2015–2462; Directorate Identifier 2014–NM–224–AD.

#### (a) Effective Date

This AD is effective June 20, 2016.

#### (b) Affected ADs

None.

#### (c) Applicability

(1) This AD applies to The Boeing Company Model 737–100, –200, –200C, –300, –400, and –500 series airplanes, certificated in any category, as identified in Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014.

(2) Installation of Supplemental Type Certificate (STC) ST01219SE ([http://rgl.faa.gov/Regulatory\\_and\\_Guidance\\_Library/rgstc.nsf/0/be866b732f6cf31086257b9700692796/\\$FILE/ST01219SE.pdf](http://rgl.faa.gov/Regulatory_and_Guidance_Library/rgstc.nsf/0/be866b732f6cf31086257b9700692796/$FILE/ST01219SE.pdf)) does not affect the ability to accomplish the actions required by this AD. Therefore, for airplanes on which STC ST01219SE is installed, a "change in product" alternative method of compliance (AMOC) approval request is not necessary to comply with the requirements of 14 CFR 39.17.

#### (d) Subject

Air Transport Association (ATA) of America Code 53, Fuselage.

#### (e) Unsafe Condition

This AD was prompted by reports of cracked antenna support channels, skin cracking underneath the number 2 VHF antenna, and cracking in the frames attached to the internal support structure. We are issuing this AD to detect and correct skin cracking of the fuselage. Such cracking could result in separation of the number 2 VHF antenna from the airplane and rapid depressurization of the cabin.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

#### (g) Inspection and Follow-On Actions: Group 1

For airplanes identified as Group 1 in Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014: Within 120 days after the effective date of this AD, inspect for cracking at the number 2 VHF antenna location, and do all applicable follow-on actions, using a method approved in accordance with the procedures specified in paragraph (m) of this AD.

#### (h) Inspection and Follow-On Actions: Groups 2 Through 6, Configurations 1 Through 3

For airplanes identified as Groups 2 through 6, Configurations 1 through 3 in Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014: Within 1,250 flight cycles after the effective date of this AD, do an external detailed inspection for cracking of the fuselage skin, as applicable, and do all

applicable corrective actions, in accordance with Part 1 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014. Thereafter, at the applicable time specified in paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014, except as required by paragraph (l)(1) of this AD: Do all applicable actions specified in paragraphs (h)(1) through (h)(4) of this AD.

(1) Repeat the Part 1 inspections specified in paragraph (h) of this AD until the accomplishment of paragraph (k)(1), (k)(2), or (k)(3) of this AD, as applicable.

(2) Inspect for cracking at the number 2 VHF antenna location using internal and external detailed inspections, internal and external high frequency eddy current (HFEC) inspections, and an HFEC open-hole inspection, as applicable, in accordance with Part 2 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014. Repeat the inspections until the accomplishment of paragraph (k)(2) or (k)(3) of this AD, as applicable.

(3) Repair any crack found, in accordance with Part 3 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014, except as required by paragraph (l)(2) of this AD.

(4) Do a preventive modification, in accordance with Part 4 of the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014, except as specified in paragraph (l)(2) of this AD. The accomplishment of this preventive modification terminates the inspections required by paragraphs (h), (h)(1), and (h)(2) of this AD.

#### (i) Inspection and Follow-On Actions: Groups 3 Through 6, Configuration 4

For airplanes identified as Groups 3 through 6, Configuration 4, in Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014: At the applicable time specified in table 10 of paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014, except as required by paragraph (l)(1) of this AD, do an external detailed inspection for cracking at the outer row of fasteners common to the internal repair doubler, and do an internal general visual inspection for cracking on the modified internal support structure of the number 2 VHF antenna, skin, and surrounding stringers, channel, and frames, in accordance with the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014.

(1) If any cracking is found, before further flight, repair using a method approved in accordance with the procedures specified in paragraph (m) of this AD.

(2) If no cracking is found, repeat the inspections at the time specified in table 10 of paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 737–53–1159, Revision 1, dated October 20, 2014.

**(j) Post Repair/Post Modification Inspections**

Tables 7 through 9 of paragraph 1.E., "Compliance," of Boeing Special Attention Service Bulletin 737-53-1159, Revision 1, dated October 20, 2014, specify post-repair and post-modification airworthiness limitation inspections in compliance with 14 CFR 25.571(a)(3) at the repaired and modified locations, which support compliance with 14 CFR 121.1109(c)(2) or 129.109(b)(2). As airworthiness limitations, these inspections are required by maintenance and operational rules. It is therefore unnecessary to mandate them in this AD. Deviations from these inspections require FAA approval, but do not require an alternative method of compliance.

**(k) Terminating Action Provisions**

The following describes terminating action for the airplane groups and configurations, as identified in Boeing Special Attention Service Bulletin 737-53-1159, Revision 1, dated October 20, 2014.

(1) For airplanes in Group 2, Configuration 2; and Groups 3 through 6, Configuration 2: Accomplishment of the inspections specified in paragraph (h)(2) of this AD terminates the repetitive inspection requirements of paragraph (h)(1) of this AD.

(2) For airplanes in Group 2, Configuration 1; and Groups 3 through 6, Configurations 1, 2, and 3: Accomplishment of the repair specified in paragraph (h)(3) of this AD terminates the initial and repetitive inspections specified in paragraphs (h), (h)(1), and (h)(2) of this AD.

(3) For airplanes in Group 2, Configuration 1; and Groups 3 through 6, Configurations 1 and 3: Accomplishment of the preventive modification specified in paragraph (h)(4) of this AD terminates the initial and repetitive inspections specified in paragraphs (h), (h)(1), and (h)(2) of this AD.

**(l) Exception to Service Bulletin Specifications**

(1) Where Boeing Special Attention Service Bulletin 737-53-1159, Revision 1, dated October 20, 2014, specifies a compliance time "after the Revision 1 date of this service bulletin," this AD requires compliance within the specified compliance time after the effective date of this AD.

(2) Where Boeing Special Attention Service Bulletin 737-53-1159, Revision 1, dated October 20, 2014, specifies to contact Boeing for appropriate action, and specifies that action as "RC" (Required for Compliance): Before further flight, repair the cracking using a method approved in accordance with the procedures specified in paragraph (m) of this AD.

**(m) Alternative Methods of Compliance (AMOCs)**

(1) The Manager, Los Angeles Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in

paragraph (n) of this AD. Information may be emailed to: [9-ANM-LAACO-AMOC-REQUESTS@faa.gov](mailto:9-ANM-LAACO-AMOC-REQUESTS@faa.gov).

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Los Angeles ACO, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) Except as required by paragraph (l)(2) of this AD, for service information that contains steps that are labeled as Required for Compliance (RC), the provisions of paragraphs (m)(4)(i) and (m)(4)(ii) apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

**(n) Related Information**

For more information about this AD, contact Jennifer Tsakoumakis, Aerospace Engineer, Airframe Branch, ANM-120L, FAA, Los Angeles ACO, 3960 Paramount Boulevard, Lakewood, CA 90712-4137; phone: 562-627-5264; fax: 562-627-5210; email: [jennifer.tsakoumakis@faa.gov](mailto:jennifer.tsakoumakis@faa.gov).

**(o) Material Incorporated by Reference**

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Boeing Special Attention Service Bulletin 737-53-1159, Revision 1, dated October 20, 2014.

(ii) Reserved.

(3) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the

National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on May 4, 2016.

**Michael Kaszycki,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2016-11200 Filed 5-13-16; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. FAA-2015-3141; Directorate Identifier 2014-NM-242-AD; Amendment 39-18516; AD 2016-10-05]**

**RIN 2120-AA64**

**Airworthiness Directives; The Boeing Company Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for all The Boeing Company Model 757 airplanes. This AD was prompted by a report of cracking in the fuselage frame. This AD requires inspections for cracking in the fuselage frame, left and right sides, and repair if necessary. We are issuing this AD to detect and correct fuselage frame fatigue cracking. Such cracking could result in loss of structural integrity and the inability to sustain loading conditions.

**DATES:** This AD is effective June 20, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of June 20, 2016.

**ADDRESSES:** For service information identified in this final rule, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, WA 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; Internet <https://www.myboeingfleet.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-3141.