

electronic access to associated maps, geospatial data and/or survey data), potential issues, and identification of constraints by Federal entities and Non-Federal entities for the proposed qualifying project;

(vii) DOE shall recommend that participating Federal entities use the Final IIP Resources Report to inform the NEPA process for the proposed qualifying project. For example, Federal entities could use the Final IIP Resources Report during scoping for an EIS and identifying potential routes, to explain why certain alternatives were eliminated from further consideration, and to preliminarily identify impacts, potential avoidance, minimization, and conservation measures, such as compensatory mitigation (onsite and offsite), developed through the use of a Regional Mitigation Approach or, where available, Regional Mitigation Strategies or Plans and considered by the project proponent to reduce the potential impacts of the proposed qualifying project to resources requiring mitigation; and

(viii) All participating Federal and Non-Federal entities shall identify a preliminary schedule for authorizations for the proposed qualifying project contingent upon timely filing of applications and related materials by the project proponent.

#### **§ 900.5 Selection of the NEPA lead agency.**

DOE, in consultation with the Federal entities, shall coordinate the selection of a potential NEPA Lead Agency responsible for preparing an environmental review document under NEPA for proposed qualifying projects. Determination and responsibilities of the NEPA Lead Agency for preparing the EIS shall be in compliance with applicable law, including the National Environmental Policy Act of 1969 and CEQ implementing regulations at 40 CFR part 1500, and each agency's respective NEPA implementing regulations and procedures. However:

(a) For proposed qualifying projects that cross lands administered by both DOI and USDA, DOI and USDA shall consult and jointly determine within thirty (30) calendar days of receiving the initiation request information from DOE which Department has a greater land management interest in the proposed qualifying project and which Department should therefore assume the role of NEPA Lead Agency.

(b) DOI and USDA shall notify DOE of their determination regarding the NEPA Lead Agency in writing within thirty (30) calendar days of making the determination.

(c) Unless DOE notifies DOI and USDA in writing of its objection to that determination within ten (10) calendar days of the DOI/USDA notification, the determination shall be deemed accepted and final. In deciding whether to object to the determination, DOE shall consider the CEQ regulations pertaining to selection of the Lead Agency, including 40 CFR 1501.5(c).

(d) For proposed qualifying projects that do not cross lands administered by both DOI and USDA, DOE and the Federal entities that will likely constitute the cooperating agencies for an environmental review document under NEPA, shall consult and jointly recommend a potential NEPA Lead Agency within 45 calendar days of receiving an IIP Process Close-Out Meeting Request. If DOE and the Federal entities are unable to agree on a recommendation for a NEPA Lead Agency, the Federal entities shall request CEQ to make a final determination by the Close-Out Meeting. No determination of a Federal entity as the potential NEPA Lead Agency under this part shall be made absent that Federal entity's consent.

#### **§ 900.6 IIP Process administrative file.**

(a) When communicating with the project proponent during the IIP Process, Federal entities are expected to include DOE in all communications related to the IIP Process for the project proponent's proposed qualifying project.

(b) DOE shall maintain all information, including documents and communications, it disseminates or receives from the project proponent, Federal entities, and Non-Federal entities during the IIP Process in an IIP Process Administrative File for future use in reviewing any applications for required Federal authorizations for the proposed qualifying project. DOE will process any requests for information from the public in accordance with Freedom of Information Act requirements. DOE will share the IIP Process Administrative File with the selected or potential NEPA Lead Agency.

(c) DOE shall document the list of issues identified during the IIP Process for a proposed qualifying project and any updates to information provided as part of the Close-Out Meeting discussion in a Final IIP Resources Report for the IIP Process Administrative File.

(d) Each Federal entity is strongly encouraged to maintain the documents and communications developed in the IIP Process subject to each Federal entity's administrative record policies

and, as appropriate and applicable, those documents and communications should become part of that Federal entity's administrative record for granting or denying a Federal authorization for each qualifying project.

[FR Doc. 2016-23285 Filed 9-27-16; 8:45 am]

BILLING CODE 6450-01-P

---

## **DEPARTMENT OF TRANSPORTATION**

### **Federal Aviation Administration**

#### **14 CFR Part 39**

**[Docket No. FAA-2016-9114; Directorate Identifier 2016-NM-146-AD; Amendment 39-18671; AD 2016-20-05]**

**RIN 2120-AA64**

#### **Airworthiness Directives; Saab AB, Saab Aeronautics (Formerly Known as Saab AB, Saab Aerosystems) Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule; request for comments.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for all Saab AB, Saab Aeronautics Model SAAB 2000 airplanes. This AD requires an inspection to identify the type of fasteners installed on the upper longerons and upper fittings of the engine mounting structure (EMS), an inspection for discrepancies of certain fasteners, and corrective action if necessary. This AD was prompted by the discovery of blind fasteners installed in EMS upper fittings that do not meet the type design. We are issuing this AD to detect and correct discrepancies of blind fasteners that could cause crack development and vibration in the engine mount structure, which could lead to failure of the affected engine-mount-to-airplane structural connection and resultant detachment of an engine from the airplane when both sides of a nacelle are affected.

**DATES:** This AD becomes effective October 13, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publications listed in this AD as of October 13, 2016.

We must receive comments on this AD by November 14, 2016.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

• *Federal eRulemaking Portal*: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

• *Fax*: 202-493-2251.

• *Mail*: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

• *Hand Delivery*: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this final rule, contact Saab AB, Saab Aeronautics, SE-581 88, Linköping, Sweden; telephone +46 13 18 5591; fax +46 13 18 4874; email [saab2000.techsupport@saabgroup.com](mailto:saab2000.techsupport@saabgroup.com); Internet <http://www.saabgroup.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-9114.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-9114; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Shahram Daneshmandi, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1112; fax 425-227-1149.

#### SUPPLEMENTARY INFORMATION:

##### Discussion

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2016-0171, dated August 22, 2016 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct

an unsafe condition for all Saab AB, Saab Aeronautics Model SAAB 2000 airplanes. The MCAI states:

During inspections, blind fasteners were found installed in engine mounting structure (EMS) upper fittings, frame NS204.7 and upper longerons. The type design specifies that the fasteners at this location must be Hi-Lok fasteners and two solid rivets (monel).

This condition, if not detected and corrected, could cause cracks development, vibration in the engine mount structure, leading to failure of the affected engine mount-to-airplane structural connection, possibly resulting in detachment of an engine from the aeroplane when affecting both sides of a nacelle.

To address this unsafe condition, SAAB issued Service Bulletin (SB) 2000-54-035 (hereafter referred to as ‘the SB’ in this [EASA] AD) to provide inspection instructions.

For the reasons described above, this [EASA] AD requires a one-time \* \* \* [general] visual inspection of the affected areas to determine which type(s) of fasteners are installed, and, depending on findings, accomplishment of applicable corrective action(s) [repair or additional actions as applicable]. This [EASA] AD also requires reporting of all inspection results to SAAB.

This [EASA] AD is considered an interim action and further AD action may follow.

Required actions include a detailed inspection for discrepancies, including gaps between the fastener head and structure, traces of movement, and deformation of the structure. You may examine the MCAI on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-9114.

#### Related Service Information Under 1 CFR Part 51

We reviewed SAAB 2000 Service Bulletin 2000-54-035, Revision 01, dated August 12, 2016. The service information describes procedures for an inspection to identify the type of fasteners installed on the upper longerons and upper fittings of the EMS, and a detailed inspection of incorrect (blind) fasteners to detect discrepancies, and corrective actions. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

#### FAA’s Determination and Requirements of This AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information

referenced above. We are issuing this AD because we evaluated all pertinent information and determined the unsafe condition exists and is likely to exist or develop on other products of the same type design.

#### FAA’s Determination of the Effective Date

An unsafe condition exists that requires the immediate adoption of this AD. The FAA has found that the risk to the flying public justifies waiving notice and comment prior to adoption of this rule because discrepancies of blind fasteners could cause crack development and vibration in the engine mount structure, which could lead to failure of the affected engine-mount-to-airplane structural connection and resultant detachment of an engine from the airplane when both sides of a nacelle are affected. Therefore, we determined that notice and opportunity for public comment before issuing this AD are impracticable and that good cause exists for making this amendment effective in fewer than 30 days.

#### Comments Invited

This AD is a final rule that involves requirements affecting flight safety, and we did not precede it by notice and opportunity for public comment. We invite you to send any written relevant data, views, or arguments about this AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA-2016-9114; Directorate Identifier 2016-NM-146-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this AD. We will consider all comments received by the closing date and may amend this AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this AD.

#### Costs of Compliance

We estimate that this AD affects 8 airplanes of U.S. registry.

We also estimate that it will take about 4 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour. Based on these figures, we estimate the cost of this AD on U.S. operators to be \$2,720, or \$340 per product.

We have received no definitive data that would enable us to provide cost

estimates for the on-condition actions specified in this AD.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

#### Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**2016–20–05 Saab AB, Saab Aeronautics (Formerly Known as Saab AB, Saab Aerosystems):** Amendment 39–18671; Docket No. FAA–2016–9114; Directorate Identifier 2016–NM–146–AD.

#### (a) Effective Date

This AD becomes effective October 13, 2016.

#### (b) Affected ADs

None.

#### (c) Applicability

This AD applies to all Saab AB, Saab Aeronautics (formerly known as Saab AB, Saab Aerosystems) Model SAAB 2000 airplanes, certificated in any category.

#### (d) Subject

Air Transport Association (ATA) of America Code 54, Nacelles/pylons.

#### (e) Reason

This AD was prompted by the discovery of blind fasteners installed in engine mounting structure (EMS) upper fittings that do not meet the type design. We are issuing this AD to detect and correct discrepancies of blind fasteners that could cause crack development and vibration in the engine mount structure, which could lead to failure of the affected engine-mount-to-airplane structural connection and resultant detachment of an engine from the airplane when both sides of a nacelle are affected.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

#### (g) Fastener Identification

Within 30 days or 150 flight hours after the effective date of this AD, whichever occurs first, do a general visual inspection of the upper longerons and upper fittings of the EMS to identify the type of fasteners installed, in accordance with the Accomplishment Instructions of SAAB 2000 Service Bulletin 2000–54–035, Revision 01, dated August 12, 2016.

#### (h) Inspection for Discrepancies

For any fastener other than the fasteners specified in SAAB 2000 Service Bulletin 2000–54–035, Revision 01, dated August 12, 2016, found during the inspection required by paragraph (g) of this AD: Before further flight, do a detailed inspection for discrepancies of those fasteners, including gaps between the fastener heads and structure, traces of movement, and deformation of the structure, in accordance with the Accomplishment Instructions of SAAB 2000 Service Bulletin 2000–54–035, Revision 01, dated August 12, 2016.

#### (i) Corrective Action

(1) If, during the inspection as required by paragraph (h) of this AD, any gap between

the fastener heads and structure, traces of movement, or deformation of the structure is found: Before further flight obtain repair instructions from the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Saab AB, Saab Aeronautics' EASA Design Organization Approval (DOA); and before further flight accomplish those instructions accordingly.

(2) If all fasteners inspected as required by paragraph (h) of this AD are firmly attached, and no deformation of the structure is found: Within 30 days or 150 flight hours after the effective date of the AD, whichever occurs first, obtain repair instructions from the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or EASA; or Saab AB, Saab Aeronautics' EASA DOA; and at the applicable time required in the repair instructions, accomplish the repair accordingly.

#### (j) Credit for Previous Actions

This paragraph provides credit for actions required by paragraphs (g) and (h) of this AD, if those actions were performed before the effective date of this AD using SAAB 2000 Service Bulletin 2000–54–035, dated July 22, 2016.

#### (k) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, International Branch, ANM–116, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Shahram Daneshmandi, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1112; fax 425–227–1149. Information may be emailed to: [9-ANM-116-AMOC-REQUESTS@faa.gov](mailto:9-ANM-116-AMOC-REQUESTS@faa.gov). Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(2) *Contacting the Manufacturer:* For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or EASA; or Saab AB, Saab Aeronautics' EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

#### (l) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2016–0171, dated August 22, 2016, for related information. You may examine the MCAI on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2016–9114.

(2) Service information identified in this AD that is not incorporated by reference is

available at the addresses specified in paragraphs (m)(3) and (m)(4) of this AD.

**(m) Material Incorporated by Reference**

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) SAAB 2000 Service Bulletin 2000-54-035, Revision 01, dated August 12, 2016.

(ii) Reserved.

(3) For service information identified in this AD, contact Saab AB, Saab Aeronautics, SE-581 88, Linköping, Sweden; telephone +46 13 18 5591; fax +46 13 18 4874; email [saab2000.techsupport@saabgroup.com](mailto:saab2000.techsupport@saabgroup.com); Internet <http://www.saabgroup.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on September 14, 2016.

**Michael Kaszycki,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2016-23081 Filed 9-27-16; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**

**[Docket No. FAA-2016-6148; Directorate Identifier 2015-NM-154-AD; Amendment 39-18660; AD 2016-19-11]**

**RIN 2120-AA64**

**Airworthiness Directives; Bombardier, Inc. Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for certain Bombardier, Inc. Model DHC-8-400 series airplanes. This AD was prompted by a malfunctioning No. 2 engine intake heater with corrosion on the thermostats and the fuselage skin where the thermostats made contact with the aircraft fuselage skin. This AD requires a general visual inspection for corrosion of the thermostats' mounting surfaces

and fuselage skin surface, corrective actions if necessary, and relocating the existing thermostats. We are issuing this AD to prevent corrosion within the thermostats that might cause the switch mechanism to seize in the open position and prevent the activation of the associated engine air intake heater. An inactive engine air intake heater could lead to an engine failure.

**DATES:** This AD is effective November 2, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of November 2, 2016.

**ADDRESSES:** For service information identified in this final rule, contact Bombardier, Inc., Q-Series Technical Help Desk, 123 Garratt Boulevard, Toronto, Ontario M3K 1Y5, Canada; telephone: 416-375-4000; fax: 416-375-4539; email: [thd.qseries@aero.bombardier.com](mailto:thd.qseries@aero.bombardier.com); Internet: <http://www.bombardier.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-6148.

**Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-6148; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Assata Dessaline, Aerospace Engineer, Avionics and Services Branch, ANE-172, FAA, New York Aircraft Certification Office (ACO), 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; telephone 516-228-7301; fax 516-794-5531.

**SUPPLEMENTARY INFORMATION:**

**Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR

part 39 by adding an AD that would apply to certain Bombardier, Inc. Model DHC-8-400 series airplanes. The NPRM published in the **Federal Register** on May 2, 2016 (81 FR 26176) ("the NPRM").

Transport Canada Civil Aviation (TCCA), which is the aviation authority for Canada, has issued Canadian Airworthiness Directive CF-2015-24, dated August 24, 2015 (referred to after this as the Mandatory Continuing Airworthiness Information, or "the MCAI"), to correct an unsafe condition for certain Bombardier, Inc. Model DHC-8-400 series airplanes. The MCAI states:

A malfunctioning Engine Air Intake Heater has been discovered with corrosion on the thermostats and the aeroplane skin where the thermostats are installed. The two thermostats are installed directly under the flight compartment floor along the aeroplane centre line where moisture accumulation and/or migration may occur, which can cause corrosion of the thermostats. Corrosion within the thermostats may seize the switch mechanism open, preventing the activation of the associated Engine Air Intake Heater. Failure of the Engine Air Intake Heater to activate may pose a safety risk to the aeroplane in icing conditions.

Bombardier has issued Service Bulletin (SB) 84-30-10 to inspect, replace if required and relocate the thermostat assembly to rectify this problem. [An inactive engine air intake heater could lead to an engine failure.]

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-6148.

**Comments**

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

**Conclusion**

We reviewed the relevant data and determined that air safety and the public interest require adopting this AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

**Related Service Information Under 14 CFR Part 51**

Bombardier, Inc. has issued Bombardier Service Bulletin 84-30-10, Revision E, dated October 10, 2014. The service information describes