

service after July 29, 2016, whichever occurs later.

(ii) Thereafter, perform a USI of the stage 8 aft web upper face every 500 cycles since last inspection.

(iii) Compliance with paragraph (f)(2)(i) of this AD is terminating action for the initial and repetitive USIs specified by paragraphs (f)(1)(i) and (f)(1)(ii) of this AD.

FIGURE 1 TO PARAGRAPH (f)—HPC STAGE 8–10 SPOOL S/Ns

Part Nos.	Serial Nos.					
1844M90G01	GWN005MF	GWNBK753	GWNBS077	GWNBS497	GWNBS724	
	GWN005MG	GWNBK754	GWNBS078	GWNBS499	GWNBS794	
	GWN0087M	GWNBK841	GWNBS079	GWNBS500	GWNBS810	
	GWN0087N	GWNBK842	GWNBS080	GWNBS501	GWNBS811	
	GWN00DGK	GWNBK843	GWNBS081	GWNBS502	GWNBS812	
	GWN00DGL	GWNBK844	GWNBS157	GWNBS609	GWNBS813	
	GWNB9J92	GWNBK952	GWNBS158	GWNBS610	GWNBS814	
	GWNBK667	GWNBK953	GWNBS159	GWNBS611	GWNBS910	
	GWNBK674	GWNBK954	GWNBS160	GWNBS612	GWNBS911	
	GWNBK675	GWNBK955	GWNBS266	GWNBS613	GWNBS912	
	GWNBK743	GWNBK956	GWNBS267	GWNBS614	GWNBS914	
	GWNBK744	GWNBK957	GWNBS268	GWNBS721	GWNBS915	
	GWNBK751	GWNBK958	GWNBS269	GWNBS722	GWNBS982	
	GWNBK752	GWNBK959	GWNBS270	GWNBS723	GWNBS983	
	1844M90G02	GWN00C2T	GWN01C5N	GWN02N8D	GWN03RTM	GWN04E21
		GWN00C2V	GWN01GE2	GWN02T3R	GWN03RTP	GWN04GHT
		GWN00G2N	GWN01GE3	GWN02WGM	GWN040RL	GWN04GHW
GWN00G2P		GWN01GE4	GWN0311K	GWN040RM	GWN04GJ0	
GWN00PPF		GWN01GE6	GWN035PP	GWN040RN	GWN04JW6	
GWN00PFR		GWN01WH1	GWN038TD	GWN040RP	GWN04JW7	
GWN00T2N		GWN02688	GWN039TG	GWN04202	GWN04JW8	
GWN00YHV		GWN02689	GWN03G2R	GWN0435W	GWN04L7K	
GWN0125G		GWN0268A	GWN03G2W	GWN04360	GWN04L7L	
GWN0125H		GWN02DP2	GWN03G30	GWN04361	GWN04MT7	
GWN0166K		GWN02DP3	GWN03JPC	GWN04362	GWN04MT8	
GWN01C5K		GWN02F9F	GWN03JPD	GWN04ATG	GWNBS984	
GWN01C5L		GWN02F9G	GWN03N8P	GWN04ATH		
GWN01C5M		GWN02L9T	GWN03N8R	GWN04E20		

(2) For all HPC stage 8–10 spools, P/N 1694M80G04, 1844M90G01, or 1844M90G02, perform an eddy current inspection (ECI) of the stage 8 aft upper face as follows:

(i) Perform an initial ECI of the stage 8 aft web upper face at the next shop visit after the effective date of this AD.

(ii) Thereafter, perform an ECI of the stage 8 aft web upper face at each subsequent shop visit.

(3) Remove from service any HPC stage 8–10 spool that fails the inspection required by paragraphs (f)(1) or (f)(2) of this AD, and replace with a spool eligible for installation.

(g) Definition

For the purpose of this AD, an engine shop visit is the induction of an engine into the shop for maintenance during which the compressor discharge pressure seal face is exposed.

(h) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: ANE-AD-AMOC@faa.gov.

(i) Related Information

(1) For more information about this AD, contact John Frost, Aerospace Engineer, Engine Certification Office, FAA, 1200 District Avenue, Burlington, MA 01803; phone: 781–238–7756; fax: 781–238–7199; email: john.frost@faa.gov.

(2) GE GE90 Service Bulletin (SB) SB 72–1151 R00, dated June 10, 2016, and Chapter 72–31–08, Special Procedures 003, and Chapter 72–00–31, Special Procedures 006, in GE GE90 Engine Manual, GEK100700, Revision 68, dated September 1, 2016, can be obtained from GE using the contact information in paragraph (i)(3) of this AD. This SB describes procedures for an on-wing USI of the stage 8 web of the stage 8–10 spool. These engine manual procedures describe how to perform ECI of the stage 8 aft web of the stage 8–10 spool.

(3) For service information identified in this proposed AD, contact General Electric Company, GE-Aviation, Room 285, 1 Neumann Way, Cincinnati, OH 45215, phone: 513–552–3272; fax: 513–552–3329; email: geae.aoc@ge.com.

(4) You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781–238–7125.

(j) Material Incorporated by Reference

None.
Issued in Burlington, Massachusetts, on November 29, 2016.

Colleen M. D'Alessandro,
Manager, Engine & Propeller Directorate,
Aircraft Certification Service.

[FR Doc. 2016–29679 Filed 12–16–16; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2016–9393; Directorate Identifier 2014–NM–199–AD]

RIN 2120–AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede Airworthiness Directive (AD) 2013–13–16, which applies to all Airbus Model A330–200, A330–200 Freighter, A330–300 series airplanes, and all Airbus Model A340–200, –300, –500, and –600 series airplanes. AD 2013–13–16 currently requires repetitive inspections for discrepancies of the ball-screw assembly of the trimmable horizontal stabilizer actuator (THSA), repetitive greasing of the THSA ball-nut, and replacement of the THSA if necessary; and modification or replacement (as applicable) of the ball-nut assembly,

which ends certain repetitive inspections. Since we issued AD 2013–13–16, we have determined that a modification that automatically detects failure of the ball-screw assembly is necessary. This proposed AD would require an inspection, corrective actions if necessary, lubrication of the ball-nut, modification of the THSA, and removal of certain airplanes from the applicability. We are proposing this AD to address the unsafe condition on these products.

DATES: We must receive comments on this proposed AD by February 2, 2017.

ADDRESSES: You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202–493–2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Airbus SAS, Airworthiness Office—EAL, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 45 80; email airworthiness.A330-A340@airbus.com; Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2016–9393; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800–647–5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Vladimir Ulyanov, Aerospace Engineer, International Branch, ANM–116,

Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1138; fax 425–227–1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA–2016–9393; Directorate Identifier 2014–NM–199–AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

On June 21, 2013, we issued AD 2013–13–16, Amendment 39–17504 (78 FR 47537, August 6, 2013) (“AD 2013–13–16”). AD 2013–13–16 requires actions intended to address an unsafe condition on all Airbus Model A330–200, A330–200 Freighter, and A330–300 series airplanes, and all Airbus Model A340–200, –300, –500, and –600 series airplanes.

Since we issued AD 2013–13–16, Airbus transferred most of the requirements of AD 2013–13–16 into airworthiness limitations, except the requirements for ECAM fault messages.

The European Aviation Safety Agency (EASA) which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2014–0219, dated September 29, 2014 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition on all Airbus Model A330 and Model A340 series airplanes. The MCAI states:

Several cases of transfer tube disconnection from the ball-nut of the trimmable horizontal stabilizer actuator (THSA) part number (P/N) 47172 and 47147–400 were detected on the ground during greasing and maintenance. Investigation results showed that this was caused by water ingress into the ball-nut, resulting in the jamming of the ball transfer circuit when the water froze. If the three (independent) ball circuits fail, then the THSA operates on a fail-safe nut (which operates without balls), which jams after several movements on the ball-screw of the THSA.

This condition, if not detected and corrected, could damage the ball-screw and the fail-safe nut, possibly resulting in jamming of the THSA and consequent reduced control of the aeroplane.

To detect at an early stage any distortion or initiation of disconnection, [Directorate General for Civil Aviation] DGAC France issued AD 2001–356 and AD 2001–357 to require repetitive inspections of the transfer tubes and their collars and, depending on findings, corrective action(s).

Prompted by another case of transfer tube disconnection, DGAC France issued AD 2001–356R2 and AD 2001–357R2 to require additional repetitive greasing and reinforcement of the ball-nut maintenance greasing instructions.

Subsequently, DGAC France issued AD 2002–037 and AD 2002–038 to require a modification that was also terminating action for the repetitive inspections and greasing tasks required by DGAC France AD 2001–356R2 and AD 2001–357R2 for the THSA P/N 47172 by application of Service Bulletin (SB) A330–27–3085 or SB A340–27–4089 (equivalent to Airbus production modification 49590), as applicable, changing the THSA P/N from 47172 to 47172–300.

Later on, DGAC France issued AD 2002–414 (later revised to R3) and AD 2002–415 (later revised to R2), which superseded the DGAC France AD 2001–356R2, AD 2001–357R2, AD 2002–037, and AD 2002–038, requiring:

- Repetitive inspections of all THSA P/N in service,
- repetitive lubrication of some THSA P/N, and
- replacement of THSA P/N 47172, 47147–400 and 47147–2XX/–3XX.

In addition, the electrical flight control computers monitor the operation of the THSA and the jamming of this actuator could be detected and indicated by messages on the maintenance system and on the [electronic centralized aircraft monitor] ECAM. For that reason, DGAC France AD 2002–414 and AD 2002–415 also required inspection of the THSA after display of such message(s).

After those [DGAC France] ADs were issued, Airbus introduced 4 new THSA, P/N 47172–500, P/N 47172–510, P/N 47172–520 and P/N 47172–530.

As these new THSA also needed to be inspected/lubricated, EASA issued [EASA] AD 2010–0192 and [EASA] AD 2010–0193, which retained the requirements of DGAC France AD F–2002–414R3 and AD F–2002–415R2 respectively, which were superseded, to add required repetitive inspections and lubrications of the new THSA P/N.

Since those [EASA] ADs were issued, all requirements of EASA AD 2010–0192 and AD 2010–0193 were transferred into Airbus Airworthiness Limitations Section (ALS) Part 4, except the requirement of paragraph (2.3) of those [EASA] ADs. At this time, compliance with ALS Part 4 tasks is required by EASA AD 2013–0268 (A330 aeroplanes) and [EASA] AD 2013–0269 (A340 aeroplanes), respectively [which correspond to FAA AD 2015–16–02, Amendment 39–18227 (80 FR 48019, August 11, 2015); and AD 2014–23–17, Amendment 39–18033 (79 FR 71304, December 2, 2014) (A340 aeroplanes); respectively.]

In addition, Airbus developed a Checkable Shear Pin (CSP) for the THSA and an associated additional electrical harness, which consists of installation of two Electrical Detection Devices (EDD) on the lower attachment secondary load path, which gives an indication to the Flight Control Primary Computers of secondary load path engagement.

After embodiment of these modifications on an aeroplane, the repetitive inspections of the ball-screw assembly for integrity of the primary and secondary load paths is no longer required, because the failure is detected automatically by this new device.

For the reasons described above, this [EASA] AD retains only the requirement of paragraph (2.3) of EASA AD 2010-0192 and 2010-0193 [actions following ECAM fault messages], which are superseded, and requires the installation of CSP and associated additional electrical harness on the THSA of the aeroplane. This [EASA] AD also requires, for A340-500/-600 aeroplanes that are post-SB A340-92-5008 (at Revision 06 or earlier), accomplishment of A340 ALS Part 3 task 274000-B0002-1-C, providing a grace period of 3 months for aeroplanes that have exceeded the applicable threshold or interval.

The unsafe condition is the degraded operation of the THSA, which could result in reduced control of the airplane.

Model A330-223F and A330-243F airplanes have been removed from the applicability of this proposed AD to correspond with the MCAI.

Required actions include a detailed inspection and corrective actions if an ECAM fault message is displayed, repetitive lubrication of the THSA ball-nut, and a modification of the THSA by installing a CSP and associated electrical harness.

Required actions also include certain "Additional Work" that is described in the following service information.

- "Additional Work" in Airbus Service Bulletin A330-27-3143, Revision 01, dated July 10, 2012, is described as removing the closing plug from the electrical harness 4515VB and connecting the electrical harness 4515VB to the THSA.

- "Additional Work" in Airbus Service Bulletin A330-92-3046, Revision 06, dated November 15, 2013; and Airbus Service Bulletin A340-92-4056, Revision 04, dated December 5, 2013; is described as replacement of a certain harness item, installation of placards and cable support, modification of a certain bracket, and installation of a certain spacer.

- "Additional Work" in Airbus Service Bulletin A340-92-5008, Revision 07, dated February 8, 2013, is described as replacing a certain wiring harness, replacing a certain THSA harness, installing additional placards, and modifying a certain wire harness installation order.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-9393.

Related Service Information Under 1 CFR Part 51

Airbus has issued the following service information.

The following service information describes procedures for doing repetitive inspections for integrity of the primary and secondary load paths of the ball-screw assembly of the THSA. These service bulletins are distinct because they apply to different airplane models.

- Airbus Service Bulletin A330-27-3102, Revision 09, dated March 29, 2016.
- Airbus Service Bulletin A340-27-4107, Revision 09, dated March 29, 2016.

The following service information describes procedures for installing two electrical detection devices, also called CSPs, on the lower attachment secondary load path of the THSA, and modifying the THSA. These service bulletins are distinct because they apply to different airplane models equipped with THS actuators having different part numbers.

- Airbus Service Bulletin A330-27-3137, dated March 20, 2007; Revision 01, dated December 6, 2007; and Revision 02, dated January 18, 2010. These service bulletins are distinct because each revision contains unique editorial changes.

- Airbus Service Bulletin A330-27-3143, Revision 01, dated July 10, 2012.
- Airbus Service Bulletin A340-27-4136, including Appendix 1, dated March 20, 2007; Revision 01, including Appendix 1, dated December 6, 2007; and Revision 02, including Appendix 1, dated February 24, 2010. These service bulletins are distinct because each revision contains unique editorial changes.

- Airbus Service Bulletin A340-27-4143, dated February 21, 2012.

- Airbus Service Bulletin A340-27-5030, Revision 01, including Appendix 1, dated November 20, 2009.

The following service information describes procedures for installing electrical wiring harnesses and brackets to connect the secondary nut detection device to the monitoring systems. These service bulletins are distinct because they apply to different airplane models.

- Airbus Service Bulletin A330-92-3046, Revision 04, dated July 15, 2010; Revision 05, dated November 7, 2011; and Revision 06, dated November 15, 2013. These service bulletins are

distinct because each revision contains unique editorial changes.

- Airbus Service Bulletin A340-92-4056, Revision 03, dated July 16, 2010; and Revision 04, dated December 5, 2013. These service bulletins are distinct because each revision contains unique editorial changes.

- Airbus Service Bulletin A340-92-5008, Revision 07, dated February 8, 2013.

This service information describes procedures for lubrication of the THSA ball-nut. These documents are distinct because they apply to different airplane models.

- Airbus A330 Airworthiness Limitations Section (ALS) Part 4—System Equipment Maintenance Requirements (SEMRs), Revision 05, dated October 19, 2015, Task 274400-00002-1-E, Lubrication of the THSA ball-nut.

- Airbus A340 ALS Part 4—SEMRs, Revision 04, dated October 19, 2015, Task 274400-00002-1-E, Lubrication of the THSA ball-nut.

- Airbus A340 ALS Part 3—Certification Maintenance Requirements (CMRs), Revision 03, dated October 19, 2015, Task 274000-B0003-1-C, Lubrication of THS Actuator Ball-screw Nut.

This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

FAA's Determination and Requirements of This Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because we evaluated all pertinent information and determined an unsafe condition exists and is likely to exist or develop on other products of the same type design.

Explanation of Proposed Compliance Times

The MCAI requires operators to modify certain airplanes by installing a CSP by a certain date. In order to provide operators with sufficient time to accomplish the modification, we have determined that a 12-month period from the effective date of this AD is acceptable. This difference had been coordinated with the EASA.

Costs of Compliance

We estimate that this proposed AD affects 33 airplanes of U.S. registry.

The actions required by AD 2013–13–16, and retained in this proposed AD take about 1 work-hour per product, at an average labor rate of \$85 per work-hour. Based on these figures, the estimated cost of the actions that are required by AD 2013–13–16 is \$85 per product.

We also estimate that it would take about 67 work-hours per product to comply with the basic requirements of this proposed AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$14,198 per product. Based on these figures, we estimate the cost of this proposed AD on U.S. operators to be \$656,469, or \$19,893 per product.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and

4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2013–13–16, Amendment 39–17504 (78 FR 47537, August 6, 2013), and adding the following new AD:

Airbus: Docket No. FAA–2016–9393; Directorate Identifier 2014–NM–199–AD.

(a) Comments Due Date

We must receive comments by February 2, 2017.

(b) Affected ADs

This AD replaces AD 2013–13–16, Amendment 39–17504 (78 FR 47537, August 6, 2013) ("AD 2013–13–16").

(c) Applicability

This AD applies to the airplanes identified in paragraphs (c)(1) and (c)(2) of this AD, certificated in any category.

(1) Airbus Model A330–201, –202, –203, –223, –243, –301, –302, –303, –321, –322, –323, –341, –342, and –343 airplanes, all manufacturer serial numbers.

(2) Airbus Model A340–211, –212, –213, –311, –312, –313, –541, and –642 airplanes, all manufacturer serial numbers.

(d) Subject

Air Transport Association (ATA) of America Code 27, Flight Controls.

(e) Reason

This AD was prompted by the determination that a modification that automatically detects failure of the ball-screw assembly is necessary. We are issuing this AD to detect and correct wear on the trimmable horizontal stabilizer actuator (THSA), possibly resulting in damage to the ball-screw and fail-safe nut, which could jam the THSA and result in reduced control of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Actions for Electronic Centralized Aircraft Monitor (ECAM) Fault Messages

For airplanes other than those identified in figure 1 to paragraphs (g), (h), and (p) of this AD: If, during any flight, one of the "PRIM X PITCH FAULT" or "STAB CTL FAULT" messages is displayed on the ECAM associated with the "PITCH TRIM ACTR (1CS)" maintenance message, before further flight after each time the message is displayed on the ECAM, do the actions specified in paragraphs (g)(1) and (g)(2) of this AD.

(1) Do the applicable detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path; check the checkable shear pins (CSP), if installed; and do all applicable corrective actions; as specified in paragraph (g)(1)(i), (g)(1)(ii), or (g)(1)(iii) of this AD. Do all applicable corrective actions before further flight.

(i) For Model A330 series airplanes: Do the actions in accordance with the Accomplishment Instructions of Airbus Service Bulletin A330–27–3102, Revision 09, dated March 29, 2016, except as required by paragraph (n)(1) of this AD.

(ii) For Model A340–200 and –300 series airplanes: Do the actions in accordance with the Accomplishment Instructions of Airbus Service Bulletin A340–27–4107, Revision 09, dated March 29, 2016, except as required by paragraph (n)(1) of this AD.

(iii) For Model A340–500 and –600 series airplanes: Do the actions using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA).

Note 1 to paragraph (g)(1)(iii) of this AD: Guidance for the inspection of the ball-screw assembly can be found in Task 274000–B0002–1–C, Inspection of the Ball-screw Assembly for Integrity of the Primary and Secondary Load Paths, of the Airbus A340 Airworthiness Limitations Section (ALS) Part 3—Certification Maintenance Requirements (CMR), Revision 03, dated October 19, 2015.

(2) Lubricate the THSA ball-nut in accordance with the applicable service information specified in paragraph (g)(2)(i), (g)(2)(ii), or (g)(2)(iii) of this AD.

(i) Task 274400–00002–1–E, Lubrication of the THSA ball-nut, of Airbus A330 ALS Part 4—System Equipment Maintenance Requirements (SEMR), Revision 05, dated October 19, 2015 (for Model A330 series airplanes).

(ii) Task 274400–00002–1–E, Lubrication of the THSA ball-nut, of Airbus A340 ALS Part 4—SEMR, Revision 04, dated October 19, 2015 (for Model A340–200 and –300 series airplanes).

(iii) Task 274000–B0003–1–C, Lubrication of THS Actuator Ball-screw Nut, of Airbus A340 ALS Part 3—CMR, Revision 03, dated October 19, 2015 (for Model A340–500 and –600 series airplanes).

FIGURE 1 TO PARAGRAPHS (g), (h), AND (p) OF THIS AD—DEFINITION OF AIRPLANE GROUPS

Group	Airplane models	On which the following actions or modifications have been done
Group 1 airplanes	Airbus Model A330–200 and –300 series airplanes.	On which the actions specified in Airbus Service Bulletin A330–27–3137, dated March 20, 2007, or Revision 01, dated December 6, 2007; and Airbus Service Bulletin A330–92–3046, Revision 04, dated July 16, 2010, or Revision 05, dated November 7, 2011; have been embodied in service.
	Airbus Model A340–200 and –300 series airplanes.	On which the actions specified in Airbus Service Bulletin A340–27–4136, including Appendix 1, dated March 20, 2007, or Revision 01, including Appendix 1, dated December 6, 2007; and Airbus Service Bulletin A340–92–4056, Revision 03, dated July 16, 2010; have been embodied in service.
Group 2 airplanes	Airbus Model A330–200 and –300 series airplanes and Model A340–200 and –300 series airplanes.	On which Airbus Modifications 55780, 52269, and 56056 have been embodied in production.
	Airbus Model A340–500 and –600 series airplanes.	On which Airbus Modifications 54882, 52191, and 56058 have been embodied in production.
Group 3 airplanes	Airbus Model A330–200 and –300 series airplanes.	On which Airbus Service Bulletin A330–27–3137, dated March 20, 2007; or Revision 01, dated December 6, 2007; has been embodied in service and Airbus Modifications 52269 and 56056 have been embodied in production.
	Airbus Model A330–200 and –300 series airplanes.	On which Airbus Modification 55780 has been embodied in production and Airbus Service Bulletin A330–92–3046 Revision 04, dated July 16, 2010; or Revision 05, dated November 07, 2011 has been embodied in service.
	Airbus Model A340–200 and –300 series airplanes.	On which Airbus Service Bulletin A340–27–4136, including Appendix 1, dated March 20, 2007; or Revision 01, including Appendix 1, dated December 6, 2007; has been embodied in service and Airbus Modifications 52269 and 56056 have been embodied in production.
	Airbus Model A340–200 and –300 series airplanes.	On which Airbus Modification 55780 has been embodied in production and Airbus Service Bulletin A340–92–4056, Revision 03, dated July 16, 2010, has been embodied in service.

(h) Installation of CSP and Electrical Harness

For all airplanes, except Group 2 airplanes specified in figure 1 to paragraphs (g), (h), and (p) of this AD, and except for airplanes

identified in paragraphs (i), (j), and (n)(2) of this AD: Within 12 months after the effective date of this AD, modify the airplane by installing a CSP on the THSA and an additional electrical harness, in accordance with the Accomplishment Instructions of the

Airbus service information specified in figure 2 to paragraph (h) of this AD, as applicable to the part number of the THSA installed on the airplane, except as provided by paragraph (n)(2) of this AD.

FIGURE 2 TO PARAGRAPH (h) OF THIS AD—APPLICABLE SERVICE INFORMATION FOR MODIFICATION

THSA Part No. (P/N)	Service Bulletin for CSP installation	Service Bulletin for electrical harness installation
47172–300	Airbus Service Bulletin A330–27–3137, Revision 02, dated January 18, 2010, for Airbus Model A330–200 and –300 series airplanes; and Airbus Service Bulletin A330–27–4136, including Appendix 1, Revision 02, dated February 24, 2010, for Airbus Model A340–200 and –300 series airplanes	Airbus Service Bulletin A330–92–3046, Revision 06, dated November 15, 2013, for Airbus Model A330–200 and –300 series airplanes; and
47147–500	Airbus Service Bulletin A330–27–3143, Revision 01, dated July 10, 2012, for Airbus Model A330–200 and –300 series airplanes; and Airbus Service Bulletin A340–27–4143, dated February 21, 2012, for Airbus Model A340–200 and –300 series airplanes.	Airbus Service Bulletin A340–92–4056, Revision 04, dated December 5, 2013, for Airbus Model A340–200 and –300 series airplanes.
47175–200	Airbus Service Bulletin A340–27–5030, Revision 01, including Appendix 1, dated November 20, 2009, for Airbus Model A340–541 and –642 airplanes.	Airbus Service Bulletin A340–92–5008, Revision 07, dated February 8, 2013, for Airbus Model A340–541 and –642 airplanes.

(i) “Additional Work” on Previously Modified Airplanes

For airplanes that have already been modified (installation of CSP on the THSA and electrical harness) before the effective date of this AD in accordance with the Accomplishment Instructions of any previous revision of an Airbus service

bulletin specified in figure 2 to paragraph (h) of this AD, as applicable: Within 12 months after the effective date of this AD, do the “Additional Work” specified in, and in accordance with, the Accomplishment Instructions of the applicable Airbus service information specified in figure 2 to paragraph (h) of this AD.

(j) Installation of Electrical Harness on Airplanes Equipped With a CSP

For airplanes having one of the THSAs installed with a part number listed in figure 3 to paragraph (j) of this AD, and which have been modified by installing a CSP on the THSA as required by paragraph (h) of this AD: Within 12 months after the effective date

of this AD, inspect to determine if the electrical harness identified in the applicable Airbus service information specified in figure 3 to paragraph (j) of this AD is installed on the airplane, and if found not to be installed, modify the airplane by installing an electrical

harness, in accordance with the Accomplishment Instructions of the Airbus service information specified in figure 3 to paragraph (j) of this AD, as applicable to the part number of the THSA installed on the airplane. Airplanes having one of the THSAs

installed with a part number listed in figure 3 to paragraph (j) of this AD already have the CSP installed on the THSA, and only the electrical harness must be installed on the airplane.

FIGURE 3 TO PARAGRAPH (j) OF THIS AD—ELECTRICAL HARNESS INSTALLATION

THSA P/N	Service Information for Electrical Harness Installation
47172–500, 47172–510, 47172–520, 47172–530, 47147–700, 47147–710.	Airbus Service Bulletin A330–92–3046, Revision 06, dated November 15, 2013, for Airbus Model A330–200 and –300 series airplanes. Airbus Service Bulletin A340–92–4056, Revision 04, dated December 5, 2013, for Airbus Model A340–200 and –300 series airplanes.
47175–500, 47175–520, 47175–530	Airbus Service Bulletin A340–92–5008, Revision 07, dated February 8, 2013, for Airbus Model A340–541 and –642 airplanes.

(k) Terminating Action for Repetitive Inspections of Airbus Model A330–200 and –300 Series Airplanes

Accomplishment of a modification before the effective date of this AD in accordance with the Accomplishment Instructions of Airbus Service Bulletin A330–27–3137, dated March 20, 2007, or Revision 01, dated December 6, 2007; and Airbus Service Bulletin A330–92–3046, Revision 04, dated July 15, 2010, or Revision 05, dated November 7, 2011; terminates the repetitive inspections specified in paragraphs (k)(1) through (k)(4) of this AD. Modification of an airplane as required by this paragraph does not constitute terminating action for the actions specified in paragraph (g)(2) of this AD or the additional work specified in paragraph (i) of this AD.

(1) Task 274400–00001–1–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and check the gap at the secondary nut trunnion, of Airbus A330 ALS Part 4—SEMR, Revision 05, dated October 19, 2015.

(2) Task 274400–00001–2–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and check the CSPs, of Airbus A330 ALS Part 4—SEMR, Revision 05, dated October 19, 2015.

(3) Task 274400–00001–3–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and check the CSPs, of Airbus A330 ALS Part 4—SEMR, Revision 05, dated October 19, 2015.

(4) Task 274400–00001–4–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and check the CSPs, of Airbus A330 ALS Part 4—SEMR, Revision 05, dated October 19, 2015.

(l) Terminating Action for Repetitive Inspections of Airbus Model A340–200 and –300 Series Airplanes

Accomplishment of a modification in accordance with the Accomplishment Instructions of Airbus Service Bulletin A340–27–4143, dated February 21, 2012; and Airbus Service Bulletin A340–92–4056, Revision 03, dated July 16, 2010; terminates the actions required by paragraph (g)(1) of this AD for modified Airbus Model A340–200 and –300 series airplanes only. Modification of an airplane as specified in

this paragraph does not constitute terminating action for the actions specified in paragraph (g)(2) of this AD, or the additional work specified in paragraph (i) of this AD.

(m) Terminating Action for Repetitive Inspections of Airbus Model A340–200 and –300 Series Airplanes

Accomplishment of a modification before the effective date of this AD in accordance with the Accomplishment Instructions of Airbus Service Bulletin A340–27–4136, including Appendix 1, dated March 20, 2007, or Revision 01, including Appendix 1, dated December 6, 2007; and Airbus Service Bulletin A340–92–4056, Revision 03, dated July 16, 2010; terminates the repetitive inspections specified in paragraphs (m)(1) through (m)(4) of this AD. Modification of an airplane as required by this paragraph does not constitute terminating action for the actions specified in paragraph (g)(2) of this AD, or the additional work specified in paragraph (i) of this AD.

(1) Task 274400–00001–1–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and gap check at the secondary nut trunnion, of Airbus A340 ALS Part 4—SEMR, Revision 04, dated October 19, 2015.

(2) Task 274400–00001–2–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and CSP check, of Airbus A340 ALS Part 4—SEMR, Revision 04, dated October 19, 2015.

(3) Task 274400–00001–3–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and CSP check, of Airbus A340 ALS Part 4—SEMR, Revision 04, dated October 19, 2015.

(4) Task 274400–00001–4–E, Detailed inspection of the ball-screw assembly for integrity of the primary and secondary load path and CSP check, of A340 ALS Part 4—SEMR, Revision 04, dated October 19, 2015.

(n) Exceptions to the Actions in Certain Service Information and Paragraph (h) of This AD

(1) Where Airbus Service Bulletin A330–27–3102, Revision 09, dated March 29, 2016 (for Model A330 series airplanes); or Airbus Service Bulletin A340–27–4107, Revision 09, dated March 29, 2016 (for Model A340 series airplanes); specifies to contact Airbus for a

damage assessment: Before further flight, accomplish the required actions in accordance with the procedures specified in paragraph (s)(2) of this AD.

(2) For airplanes that already had the electrical harness installed during production using Airbus Modifications 52269 and 56056 for Airbus Model A330–200 and –300 series airplanes and Airbus Model A340–200 and –300 series airplanes, and using Airbus Modifications 52191 and 56058 for Model A340–500 and –600 series airplanes: Only the CSP must be installed on the THSA in accordance with applicable Airbus service bulletins and within the compliance time specified in paragraph (h) of this AD.

(o) Terminating Action for Repetitive Inspections for Airplanes on Which Actions Required by Paragraph (h), (i), or (j) of This AD Are Done

Modification of an airplane as required by paragraph (h), (i), or (j) of this AD, as applicable, constitutes terminating action for that airplane for the applicable actions identified in paragraphs (o)(1) through (o)(4) of this AD.

(1) For all airplanes: The actions required by paragraph (g) of this AD.

(2) For Model A340–500 and –600 series airplanes: Task 274000–B0002–1–C, Inspection of the Ball-screw Assembly for Integrity of the Primary and Secondary Load Paths, of Airbus A340 ALS Part 3—CMR, Revision 03, dated October 19, 2015.

(3) For Model A330–200 and –300 series airplanes: The ALS tasks identified in paragraphs (k)(1) through (k)(4) of this AD.

(4) For Model A340–200 and –300 series airplanes: The ALS tasks identified in paragraphs (m)(1) through (m)(4) of this AD.

(p) Ball-Screw Assembly Inspection for Certain Airplanes

For Model A340–500 and –600 airplanes that are in post-Airbus Service Bulletin A340–92–5008, at Revision 06 or earlier, configuration: Before exceeding the threshold or interval, as applicable, of Task 274000–B0002–1–C, Inspection of the Ball-screw Assembly for Integrity of the Primary and Secondary Load Paths, of Airbus A340 ALS Part 3—CMR, Revision 03, dated October 19, 2015, or within 3 months after the effective date of this AD, whichever occurs later, accomplish Task 274000–B0002–1–C, Inspection of the Ball-screw Assembly for

Integrity of the Primary and Secondary Load Paths, of Airbus A340 ALS Part 3—CMR, Revision 03, dated October 19, 2015; and do all applicable corrective actions. Do all applicable corrective actions before further flight using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the EASA; or Airbus's EASA DOA. Repeat Task 274000-B0002-1-C, Inspection of the Ball-screw Assembly for Integrity of the Primary and Secondary Load Paths, thereafter at the applicable intervals specified in Airbus A340 ALS Part 3—CMR, Revision 03, dated October 19, 2015.

(q) Parts Installation Prohibitions

(1) For all airplanes except Group 2 airplanes as identified in figure 1 to paragraphs (g), (h), and (p) of this AD: After modification of the airplane as required by paragraph (h), (i), or (j) of this AD, as applicable, no person may install any THSA having P/N 47172-300, P/N 47147-500, P/N 47175-200, or P/N 47175-300.

(2) For Group 2 airplanes, as identified in figure 1 to paragraphs (g), (h), and (p) of this AD: As of the effective date of this AD, no person may install on any Group 2 airplane any THSA having P/N 47172-300, P/N 47147-500, P/N 47175-200, or P/N 47175-300.

(r) Credit for Previous Actions

This paragraph provides credit for actions required by paragraph (g)(2) of this AD, if those actions were performed before the effective date of this AD using the applicable service information specified in paragraphs (r)(1) through (r)(4) of this AD.

(1) Task 274400-00002-1-E, Lubrication of the THSA ball-nut, of Airbus A330 ALS Part 4—Ageing Systems Maintenance, Revision 03, dated September 9, 2011 (for Model A330 series airplanes).

(2) Task 274400-00002-1-E, Lubrication of the THSA ball-nut, of Airbus A330 ALS Part 4—Ageing Systems Maintenance, Revision 04, dated August 27, 2013 (for Model A330 series airplanes).

(3) Task 274400-00002-1-E, Lubrication of the THSA ball-nut, of Airbus A340 ALS Part 4—Ageing Systems Maintenance, Revision 02, dated October 12, 2011 (for Model A340-200 and -300 series airplanes).

(4) Task 274400-00002-1-E, Lubrication of the THSA ball-nut, of Airbus A340 ALS Part 4—Ageing Systems Maintenance, Revision 03, dated November 15, 2012 (for Model A340-200 and -300 series airplanes).

(s) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Vladimir Ulyanov, Aerospace Engineer, International Branch, ANM-116, Transport

Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1138; fax 425-227-1149. Information may be emailed to: 9-ANM-116-ACO-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Contacting the Manufacturer: As of the effective date of this AD, for any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or EASA; or Airbus's EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

(t) Related Information

(1) Refer to Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2014-0219, dated September 29, 2014, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-9393.

(2) For service information identified in this AD, contact Airbus SAS, Airworthiness Office—EAL, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 45 80; email airworthiness.A330-A340@airbus.com; Internet <http://www.airbus.com>.

(3) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on November 10, 2016.

Michael Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF THE TREASURY

Internal Revenue Service

26 CFR Part 1

[REG-140328-15]

RIN 1545-BN17

Guidance Regarding Predecessors and Successors Under Section 355(e); Limitation on Gain Recognition; Guidance Under Section 355(f)

AGENCY: Internal Revenue Service (IRS), Treasury.

ACTION: Withdrawal of notice of proposed rulemaking, notice of proposed rulemaking by cross-reference to temporary regulations.

SUMMARY: In the Rules and Regulations section of this issue of the **Federal Register**, the IRS is issuing temporary regulations that provide guidance regarding the distribution by a distributing corporation of stock or securities of a controlled corporation without the recognition of income, gain, or loss. The temporary regulations provide guidance in determining whether a corporation is a predecessor or successor of a distributing or controlled corporation for purposes of the exception under section 355(e) of the Internal Revenue Code to the nonrecognition treatment afforded qualifying distributions, and they provide certain limitations on the recognition of gain in certain cases involving a predecessor of a distributing corporation. The temporary regulations also provide rules regarding the extent to which section 355(f) causes a distributing corporation (and in certain cases its shareholders) to recognize income or gain on the distribution of stock or securities of a controlled corporation. Those temporary regulations affect corporations that distribute the stock or securities of controlled corporations and their shareholders or security holders of those distributing corporations. The text of those temporary regulations serves as the text of these proposed regulations.

DATES: Comments and requests for a public hearing must be received by March 20, 2017.

ADDRESSES: Send submissions to: CC:PA:LPD:PR (REG-140328-15), Room 5203, Internal Revenue Service, P.O. Box 7604, Ben Franklin Station, Washington, DC 20044. Submissions may be hand-delivered Monday through Friday between the hours of 8 a.m. and 4 p.m. to CC:PA:LPD:PR (REG-140328-15), Courier's Desk, Internal Revenue Service, 1111 Constitution Avenue NW., Washington, DC 20224, or sent electronically, via the Federal eRulemaking Portal at <http://www.regulations.gov/> (REG-140328-15).

FOR FURTHER INFORMATION CONTACT:

Concerning the proposed regulations, Richard K. Passales at (202) 317-5024 or Marie C. Milnes-Vasquez, (202) 317-7700; concerning submission of comments, and/or requests for public hearing, Regina Johnson at (202) 317-6901 (not toll-free numbers).

SUPPLEMENTARY INFORMATION:

Background and Explanation of Provisions

On November 22, 2004, the Treasury Department and the IRS published in the **Federal Register** (69 FR 67873) a